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Instructions for Continued Airworthiness

Cargo Swing Suspension System
For the
Airbus Helicopters AS350 Series

Part Number 200-280-04

STC SR01164SE

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Record of Revisions

Revision	*Date*	*Page(s)*	*Reason for Revision*
6	09/10/15	Section 5, Pages 1, 11, 13, 14, 16 – 18	Updated definition of "hours of external load operations", clarified parts requiring NDT, added inspection criteria for 235-117-00, expanded re-assembly instructions for swing frame assembly after inspection.
7	07/20/16	Section 4, Section 5 pages 1, 6, 11 Section 25 pages 4-6, 8, 10, 18, 19, 24	Updated Airworthiness Limitations section, clarified maintenance interval for suspension assembly, clarified NDT of load cell assembly, updated electrical schematic and troubleshooting table, added bracket P/N 235-274-00 and added notices that fuel drain guard may not be present.
8	02/21/18	Section 5 pages 11, 15	Removed magnetic particle inspection requirement for load cell assembly, inserted instructions to return load cell to factory for inspection/calibration. Listed overhaul kit P/N 212-040-00 for swing suspension. Revised attach bolt diameter limit to .495" to standardize with cargo hook CMMs.
9	06/20/24	Section 5 Section 25, page 23	Added disassembly and inspection of upper load cell joint to annual/100 hour inspection. Re-formatted inspection figures and inspection table. Added tightening instructions for nut at upper load cell pivot joint.
10	10/23/24	Section 5 Pages 1, 10	Added note specifying no maintenance to be performed on the load weigh indicator. Corrected bolt P/N 511-239-00 to 510-762-00.
11	07/15/26	Section 5 pages 5 -18	Updated Figure 5.2.2 to show sub-assembly P/N 232-145-25 parts. Corrected Hook Bumper P/N to 290-774-00 and nut P/N 510-718-00 to P/N 510-440-00 (P/N 510-718-00 is optional). Listed additional P/Ns. Added Special Inspection for Torsional Loads. Updated nut P/N 510-440-00 to P/N 510-718-00, 510-440-00 is alternate.

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Section 0
Introduction

0.4 Scope

The following information is necessary to carry out the service, maintenance, and inspection of the Cargo Hook Swing Suspension System P/N 200-280-04. See section 25.2 for a detailed description of the kit.

0.5 Purpose

The purpose of this Instructions for Continued Airworthiness (ICA) manual is to provide the information necessary to inspect, service, and maintain in an airworthy condition the P/N 200-280-04 Cargo Hook Swing Suspension System.

0.6 Arrangement

This manual contains instructions for the service, maintenance, inspection and operation of the Cargo Hook Swing Suspension System P/N 200-280-04 on Airbus Helicopters Model AS350 series helicopters. The manual is arranged in the general order that maintenance personnel would use to install, maintain and operate the Cargo Hook Swing Suspension System in service.

The arrangement is:

Section 0 Introduction
Section 4 Airworthiness limitations (None apply to this System.)
Section 5 Inspection and overhaul schedule
Section 11 Placards and Markings
Section 12 Servicing
Section 25 Equipment and Furnishings

0.7 Applicability

These Instructions for Continued Airworthiness are applicable to Cargo Hook Swing Suspension System P/N 200-280-04 (with Cargo Hook P/N 528-029-00) for the Airbus Helicopters AS350 series helicopters. Refer to the appropriate Airbus Helicopters maintenance documentation for instructions regarding parts of the aircraft that interface with the P/N 200-280-04 system.

0.9 Abbreviations

FAA Federal Aviation Administration
ICA Instructions for Continued Airworthiness
CFR Code of Federal Regulations

0.12 Precautions

The following definitions apply to safety labels used in this manual.



Indicates a hazardous situation which, if not avoided, will result in death or serious injury.



Indicates a hazardous situation which, if not avoided, could result in death or serious injury.



Indicates a hazardous situation which, if not avoided, could result in minor or moderate injury.



Draws the reader's attention to important or unusual information not directly related to safety.



Used to address practices not related to personal injury.

0.19 Distribution of Instructions for Continued Airworthiness

Before performing maintenance ensure that the Instructions for Continued Airworthiness (ICA) in your possession is the most recent revision. Current revision levels of all manuals are posted on Onboard Systems Int'l web site at www.onboardsystems.com. Also a Documentation Update Service is available on the web site. Registering for this service provides an e-mail or fax notification when a manual has been revised. Hard copies of all manuals are available from the factory, contact the factory at 800-275-0883 to request a copy.

Section 4
Airworthiness Limitations

The Airworthiness Limitations section is FAA approved and specifies inspections and other maintenance required under Secs. 43.16 and 91.403 of the Federal Aviation Regulations unless an alternative program has been FAA approved.

No airworthiness limitations are associated with this type design change.

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Section 5
Inspection and Overhaul Schedule

The scheduled inspections (Annual/100 Hour and 5 Year/1000 Hour) are defined herein. If the system is subjected to unusual circumstances, extreme environmental conditions, etc., it is the responsibility of the operator to perform the inspections more frequently to ensure proper operation. Refer to Section 5.1 for the Annual/100 Hour Inspection and Section 5.2 for the 5 Year/1000 Hour Inspection.

There is no maintenance to be performed on the load weigh indicator. Do not open the enclosure, if repair is needed return it to the factory.

5.1 Annual/100 Hour Inspection

Annually, or 100 hours of external load operations, whichever comes first, inspect the cargo swing suspension per the following. A grace period of 1 month or 10 hours of external load operations can be applied to this interval for maintenance scheduling convenience only. Refer also to Component Maintenance Manual (CMM) 122-017-00 for additional inspection.

NOTICE

*Hours of external load operations should be interpreted to be **(1)** anything is attached to the primary cargo hook (whether or not a useful load is being transported) and **(2)** the aircraft is flying. If these conditions are **NOT** met, time does **NOT** need to be tracked.*

1. Activate the electrical system and press the Cargo Release button to ensure the cargo hook electrical release system is operating correctly. With no load on it, the cargo hook must release. Reset the hook by hand after release.

CAUTION

Depressing the button continuously in excess of 20 seconds will cause the cargo hook solenoid to overheat, possibly causing permanent damage.

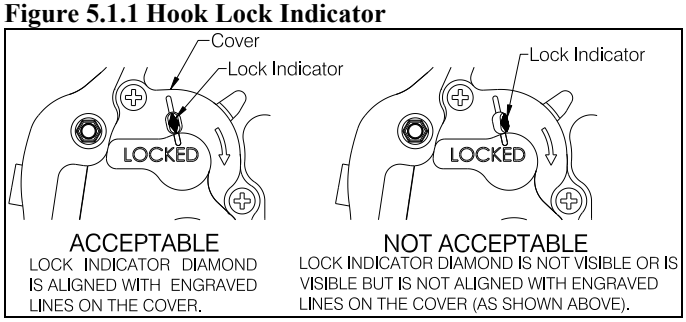
2. Activate the manual release system by pulling the release lever in the cockpit. With no load on it, the cargo hook must release. Reset the cargo hook by hand after release. Verify that the hook lock indicator on the side of the hook returns to the fully locked position.

⚠ CAUTION

In the fully locked position, the hook lock indicator must align with the lines on the manual release cover (see Figure 5.1.1).

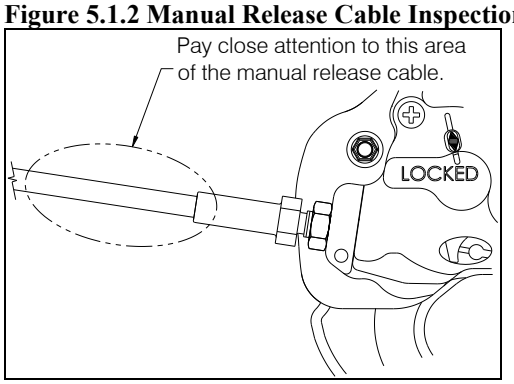
5.1 Annual/100 Hour Inspection continued

Figure 5.1.1 Hook Lock Indicator



3. Visually inspect the external manual release cable for damage and security, with emphasis on the cargo hook end of the release cable (refer to Figure 5.1.2).

Figure 5.1.2 Manual Release Cable Inspection

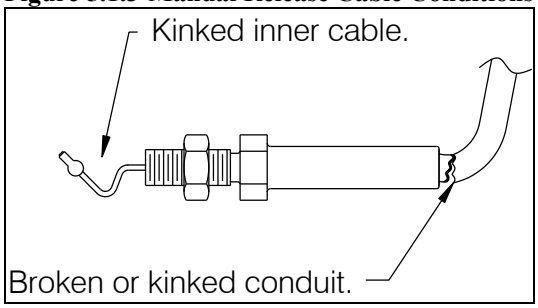


4. Remove the manual release cover from the cargo hook and inspect the visible section of the inner cable for kinks or frays.

⚠ CAUTION

Manual release cables are wearable items and must be replaced as condition requires. Broken or kinked conduit, inner cable kinks (ref Figure 5.1.3), frays, or sticky operation are each cause for immediate replacement.

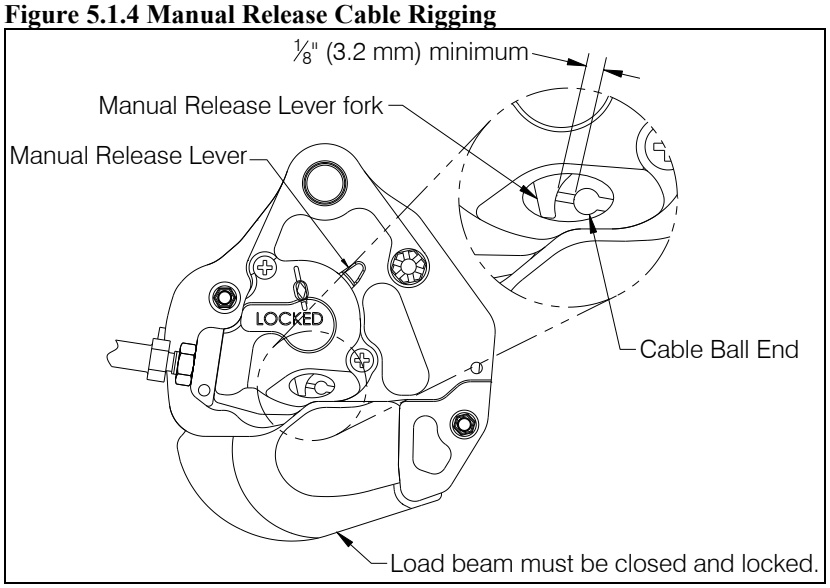
Figure 5.1.3 Manual Release Cable Conditions



5.1 Annual/100 Hour Inspection continued

5. Check the manual release cable rigging. With the cargo hook load beam closed and locked, rotate the manual release lever clockwise to remove the free play (the free play is taken up when the hook lock indicator begins to move, this is also readily felt as the lever rotates relatively easily for several degrees as the free play is taken up) and hold it in this position while checking the gap between the release lever fork and the cable ball end as shown below. A minimum gap of 1/8" (3.2 mm) should be present as shown in Figure 5.1.4.

Figure 5.1.4 Manual Release Cable Rigging



6. Visually inspect for presence and security of fasteners and electrical connections.
7. Visually inspect the external electrical wire harnesses for damage, chafing and security.
8. Visually inspect for corrosion on the exterior of the cargo hook, load cell and swing suspension components.
9. Visually inspect the cargo hook and swing frame assembly bumpers for damage and security.
10. Swing the cargo hook and the swing suspension and ensure all pivot points rotate freely without binding.
11. Move the cargo hook and the swing suspension throughout their full ranges of motion and observe the manual release cable, electrical harnesses, and ground strap to ensure that they have enough slack. The release cable, harnesses, and ground strap must not be the stops that prevent the cargo hook or suspension from moving freely in all directions.

5.1 Annual/100 Hour Inspection continued

12. Visually inspect for cracks in the welded suspension frame. Pay special attention to the areas around the welds. The frame tubes contain a corrosion preventative compound, which may leak out through a crack and provide an indication. At any sign of cracking, remove and replace discrepant part.

13. Inspect suspension cables for broken strands, paying close attention to the sections around the thimbles at each end. Pass a cloth over the cables. This will clean the load ropes for a visual inspection and detect broken wires if the cloth snags on the rope. Ten randomly distributed broken wires in one rope lay (one complete rotation around the wire) or five broken wires in one strand in one rope lay is considered unacceptable. One rope lay is the length along the rope which a single strand requires to make one complete spiral around the core.

14. Disassemble and inspect the upper joint of the load cell (16) per the following, referring to Figure 5.2.2 for item numbers in parentheses ( ). If Swing Frame Assembly has been subjected to torsional loading, see Special Inspection for Torsion involving further disassembly and inspection instructions in step 15.

    a. Remove the cotter pin (22), nut (21), washer (20), and Retaining Bushing (18) from the end of the bolt (19).

    b. While supporting the load cell and cargo hook (28), remove the bolt and the other Retaining Bushing (18) and slide the Gimbal Shaft (17) out of the joint, separating the load cell from the Gimbal (1.1).

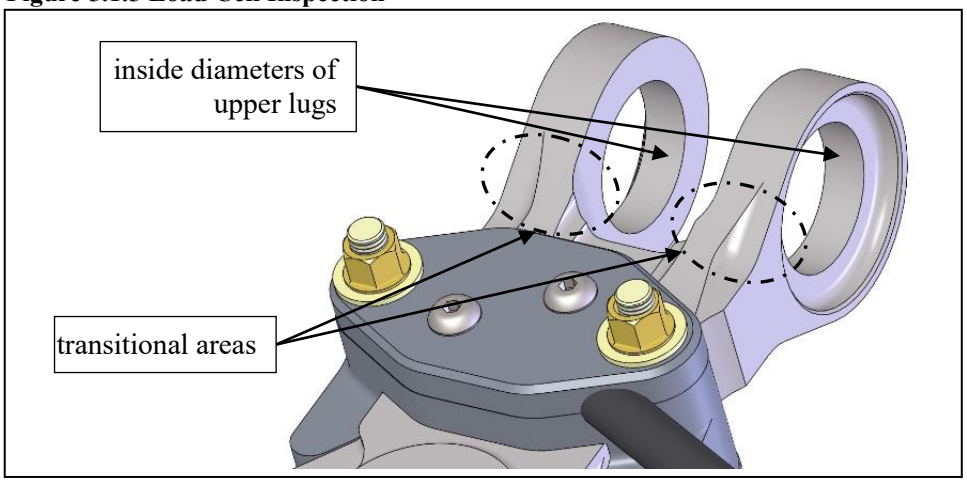
    c. Visually inspect the Gimbal Shaft for signs of wear, such as pitting, galling, or reduction in diameter. If reduction in diameter is observed, refer to the criteria in Table 5.2.2. There should be no visible wear, except for light burnishing of the outside diameter surface. If the burnishing can be removed using Scotch Brite (3M P/N 7447), it is considered light.

    d. Inspect the inside diameters of the upper lugs of the load cell (refer to Figure 5.1.5) for signs of wear such as pitting, galling, or elongation of the holes. If elongation is observed, refer to the size limits criteria in Table 5.2.2. There should be no visible wear, aside from light burnishing of the inside diameter surfaces. If the burnishing can be removed using Scotch Brite, it is considered light.

       Visually inspect for cracks around both lugs, including the transitional areas (indicated in Figure 5.1.5) where the lugs transition into the flat portion of the load cell. Inspect these areas on both sides.

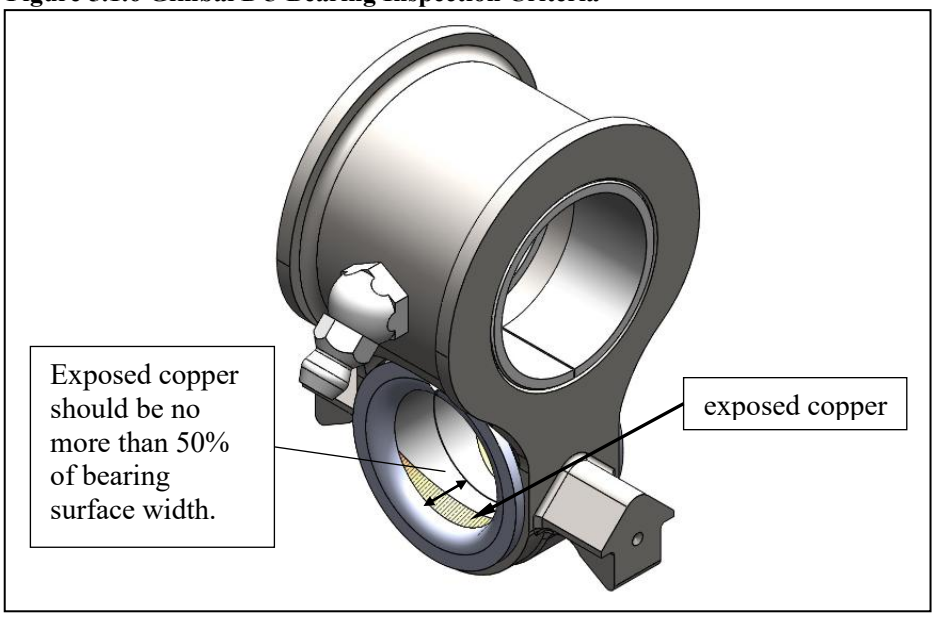
5.1 Annual/100 Hour Inspection continued

Figure 5.1.5 Load Cell Inspection



e. Inspect the flanged DU bearings (1.2) of the Gimbal (1.1) that interface with the Gimbal Shaft. As these bearings wear, the low-friction PTFE layer is removed, exposing the copper layer below (ref. Figure 5.1.6). Across the width of the bearing surface, no more than 50% should be the exposed copper layer.

Figure 5.1.6 Gimbal DU Bearing Inspection Criteria



f. Replace worn or damaged parts. If DU bearings require replacement (refer to Section 5.2 for further disassembly instructions), press in with wet zinc chromate primer (TTP-1757 or equivalent) applied to the inside diameter of the mating hole.

g. Re-assemble the load cell to the Gimbal per Section 25.17.

5.1 Annual/100 Hour Inspection continued

Special Inspection for Torsional Loads

If the swing frame assembly is used with torsional loads (such as a fertilizer slinger bucket) that are attached directly to the cargo hook via a spreader bar (without any additional restraints), perform additional inspections as specified in this below.

WARNING

Loads attached to the cargo hook with a spreader bar (such as a typical fertilizer bucket) cause the swing frame assembly to react high torsional loads. Repetitive high torsional loads are damaging to the swing frame assembly and require an additional inspection as defined in the following step.

15. Disassemble the upper pivoting joint of the swing frame assembly to remove and inspect the Gimbal Assembly per the following.

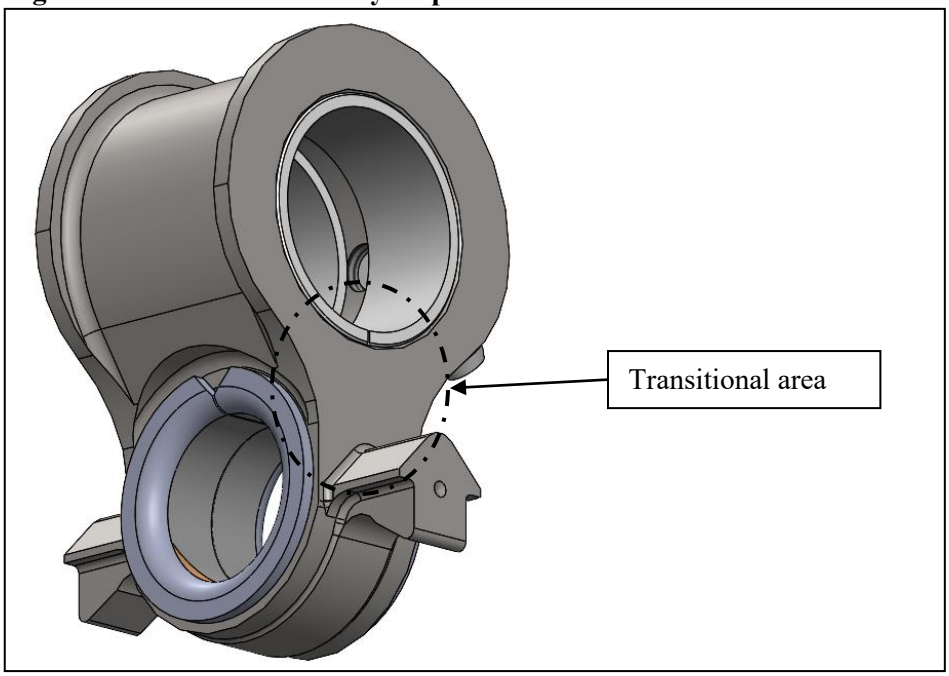
    a. Remove the cotter pin (8), nut (9), Shaft Cap (5) and remove the bolt (10) and the other Shaft Cap.

    b. Remove the Pivot Shaft while supporting each Swing Frame Half.

    c. Remove the Gimbal Assembly (1) and the Thrust Washers (15).

    d. Perform a visual inspection for cracks in the Gimbal (1.1), with specific attention on the transitional areas (on both sides) as identified below.

Figure 5.1.7 Gimbal Assembly Inspection



5.1 Annual/100 Hour Inspection continued

If the fuel drain guard is installed perform the following.

1. Inspect all fuel drain guard parts for corrosion, gouges, nicks, and dents. If depth of corrosion pits, gouges, nicks, or dents exceed .060", remove and replace damaged part(s).
2. Inspect Guard for damage that causes lever to bind or rub. Remove and replace Guard if it is obstructing free operation of lever.
3. Inspect for fuel leakage. If leakage is noted, re-apply sealant to Retainer as described in Section 25. Refer to Airbus Helicopters maintenance instructions for other possible causes.

5.2 5 Year/1000 Hour Inspection

Every 5 years or 1000 hours of external load operations, whichever comes first, remove the external components from the aircraft and disassemble per the following. A six-month or 100-hour grace period can be applied if needed for maintenance scheduling convenience only. No extension to maintenance is allowed beyond this tolerance grace period.

*After disassembly return the Load Cell to the factory for inspection and calibration. The factory will inspect the condition of the load cell and perform acceptance test procedures including calibration and zero balance, repairing as necessary.

Remove the suspension by removing the quick release pins at the cable attachments to the shackle assemblies at the landing gear cross tubes and disconnecting the electrical harnesses, ground strap and manual release cable connections at the aircraft. Remove the four Shackle Assemblies from the aircraft hardpoints.

Disassemble per the following steps. For item numbers in parentheses ( ), refer to Figure 5.2.1 through Figure 5.2.3 for Swing Suspension parts and Figure 5.2.4 for Shackle Assembly parts. Part numbers are listed in Table 5.2.1.

1. Remove the four suspension cable assemblies from the swing frame feet by removing the cotter pin (35), nut (34), two washers (33) and bolt (31). Remove the Standoff Bushings (32).
2. Separate the Load Cell (16) and Cargo Hook (28) from the Swing Frame by removing the cotter pin (22), nut (21), washer (20), and Shaft Retaining Bushing (18) and sliding the Gimbal Shaft (17) and bolt (19) out with the other Shaft Retaining Bushing (18).
3. At each foot of the swing frame remove the nut (7) and bolt (12) that secure the rod end fittings (13) to the swing frame feet.
4. At the upper pivoting joint of the swing frame remove the cotter pin (8) and nut (9) from the end of the bolt (10) and remove the Shaft Cap (5).
5. Slide the Swing Frame Half (3) off of the Pivot Shaft (4) and remove the bumper (6), thrust washers (15), and Gimbal Assembly (1.1, 1.2, 1.3).
6. Slide the Pivot Shaft out of the opposite Swing Frame Half (3) and remove the other Shaft Cap (5) and bolt.
7. Separate the Cargo Hook and Hook Bumper (23) from the Load Cell by removing the cotter pin (8), nut (27), and washers (25, 26) from the end of the Attach Bolt (24) and slide the Attach Bolt out, removing the other washer (25).
8. Separate the Gimbal Assembly (36.1, 36.2, 36.3) from each suspension cable assembly by removing the cotter pin (35), nut (34), washer (33), and bolt (39).
9. Remove spiral wrap from around the frame tubes.
10. Remove bushings if it is necessary that they be replaced. All self-lubricating bushings use an interference fit to hold them in place. Use an arbor press or similar to press the bushings out of the bore they are mounted in. Do NOT use heat.

5.2 5 Year/1000 Hour Inspection continued

Figure 5.2.1 Suspension Cable to Swing Frame Attachment Hardware

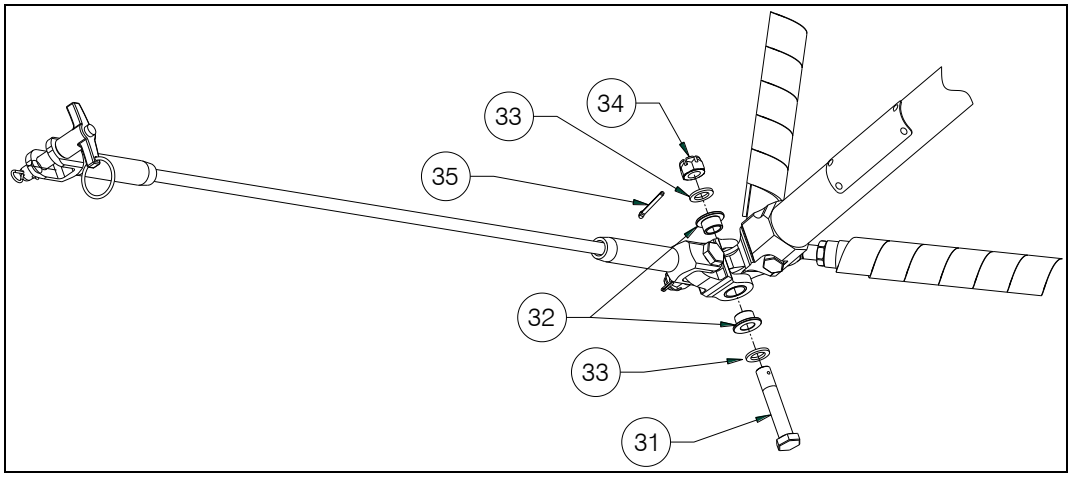
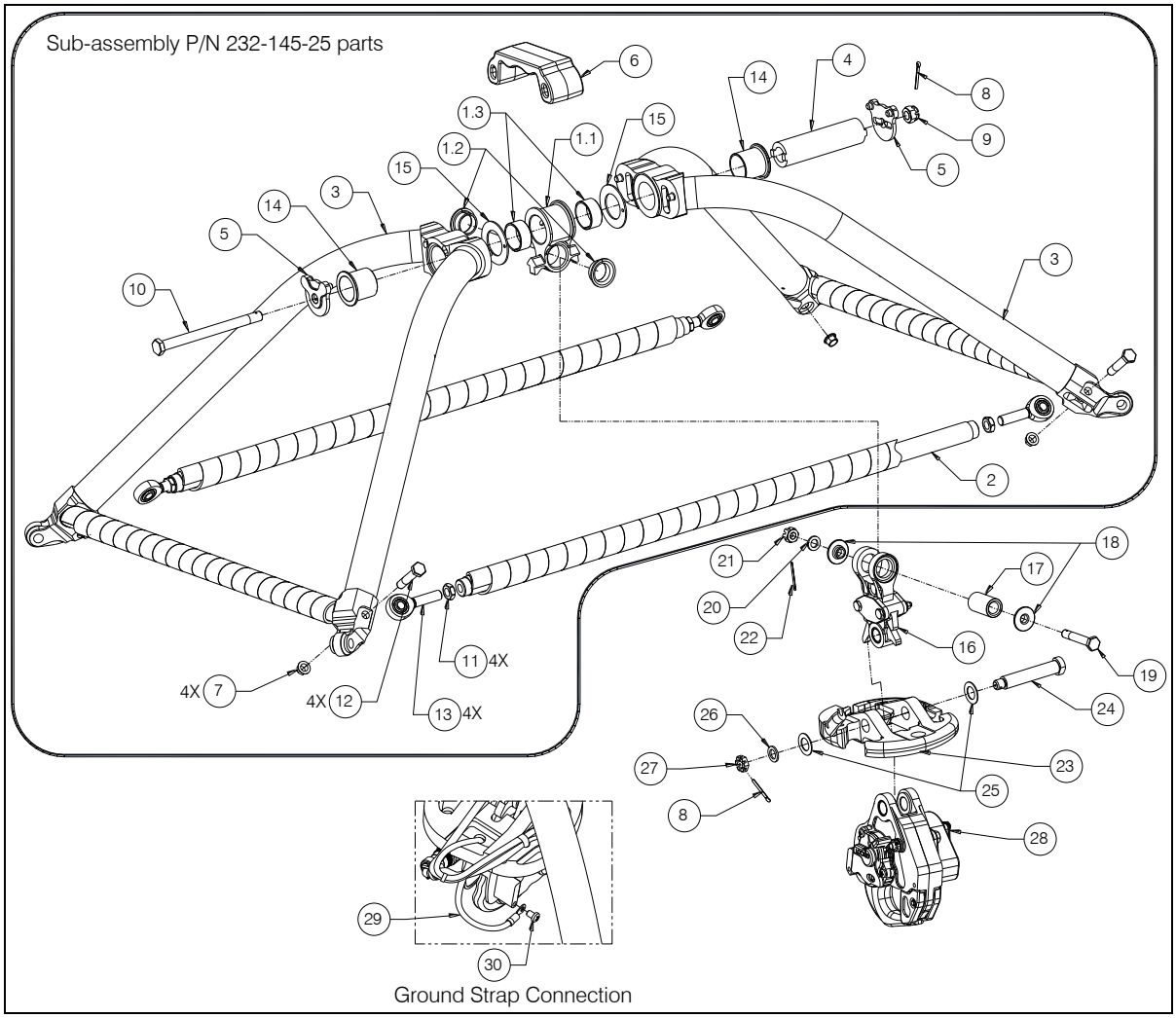


Figure 5.2.2 Swing Frame Assembly P/N 232-145-03 Parts



5.2 5 Year/1000 Hour Inspection continued

Figure 5.2.3 Suspension Cable Parts

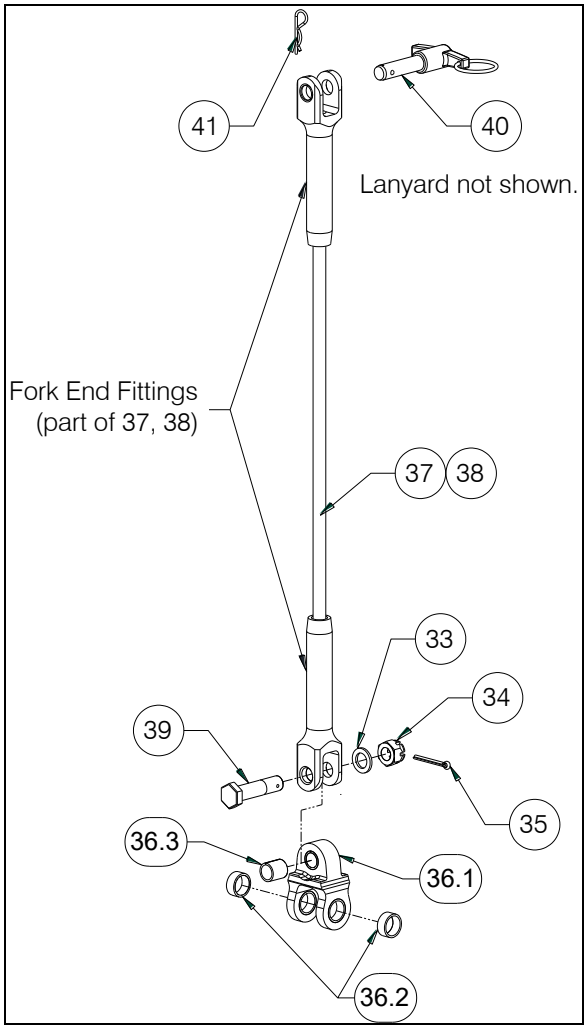
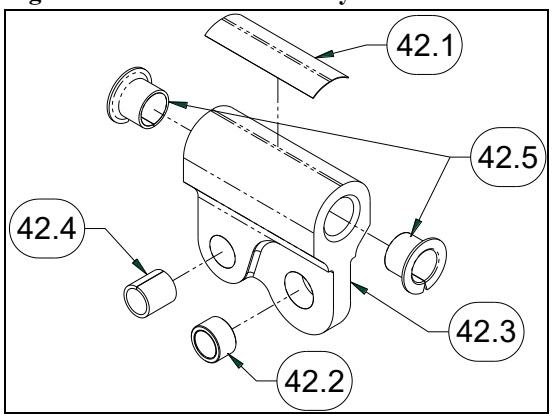


Figure 5.2.4 Shackle Assembly Parts



5.2 5 Year/1000 Hour Inspection continued

Table 5.2.1 Swing Suspension Parts List

Item	Part No.	Description	Qty
1[1]	232-143-01	Load Cell Gimbal Assembly	1
1.1	290-841-00	Gimbal	1
1.2	517-046-00	Flanged DU Bearing	2
1.3	517-056-00	DU Bearing	2
1.4[3]	518-003-00	Grease Fitting	1
2	235-116-00	Frame Strut Weldment	2
3	235-117-00	Swing Frame Half	2
4	290-842-00	Pivot Shaft	1
5	290-843-00	Cap-Shaft	2
6	290-862-00	Bumper	1
7	510-104-00	Nut	4
8	510-178-00	Cotter Pin	2
9	510-440-00	Nut	1
10	510-506-00	Bolt	1
11	510-510-00	Jam Nut	4
12	510-762-00	Bolt	4
13	517-055-00	Spherical Rod End	4
14	517-057-00	Flanged DU Bearing	2
15	517-058-00	Thrust Bearing	2
16	210-249-03	Load Cell	1
17	290-739-00	Shaft - Gimbal	1
18	290-740-00	Retaining Bushing - Shaft	2
19	510-443-00	Bolt	1
20	510-220-00	Washer	1
21	510-320-00	Nut	1
22	510-115-00	Cotter Pin	1
23	290-774-00	Hook Bumper	1
24	290-775-00	Long Attach Bolt	1
25	510-183-00	Washer	2
26	510-174-00	Washer	1
27	510-170-00	Nut	1
28	528-029-00	Cargo Hook	1
29	270-126-00	Ground Strap	1
30	510-391-00	Screw	1
31	510-439-00	Bolt	4
32	290-749-00	Standoff Bushing	8
33	510-221-00	Washer	12
34	510-718-00[4]	Nut	8
35	510-178-00	Cotter Pin	8

5.2 5 Year/1000 Hour Inspection continued

Table 5.2.1 Swing Suspension Parts List continued

Item	Part No.	Description	Qty
36[1]	232-142-00	Lower Attach Gimbal Assembly	4
36.1	290-746-00	Gimbal	1[2]
36.2	517-048-00	DU Bearing	2[2]
36.3	517-016-00	DU Bearing	1[2]
37	232-177-00	Forward Suspension Cable	2
38	232-178-00	Aft Suspension Cable	2
39	510-438-00	Bolt	4
40	290-851-00	Quick Release Pin	4
41	514-048-00	Safety Pin	4
41[1]	232-137-01	Shackle Assembly	4
42.1	215-396-00	Shackle Assembly Placard	1[2]
42.2	290-750-00	Attach Fitting Bushing	1[2]
42.3	290-850-00	Shackle	1[2]
42.4	517-016-00	DU Bearing	1[2]
42.5	517-047-00	Flanged DU Bearing	2[2]
Item(s) not shown in figures			
-	268-024-00	Manual Release Cable	1

1 Item not shown assembled.

2 Quantity is per assembly.

3 Not shown.

4 Alternate nut P/N is 510-440-00

Notes:

1) Load Cell P/N 210-249-03 supersedes P/Ns 210-249-00, 210-199-01, and 210-199-00.
2) Swing Frame Assembly P/N 232-145-25 is comprised of items 1 through 15.
3) Optionally use nut P/N 510-718-00 in place of nut P/N 510-440-00.
4) Spiral Wrap P/N 590-013-00 is wrapped around the horizontal swing frame tubes.

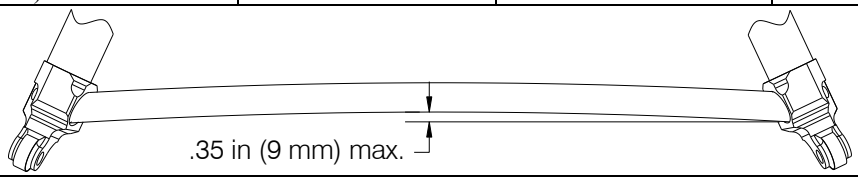
5.2 5 Year/1000 Hour Inspection continued

Return the Load Cell Assembly (P/N 210-249-03) to the factory for inspection and calibration. The factory will inspect the condition of the load cell and perform acceptance test procedures including calibration and zero balance, repairing as necessary.

In addition, carefully inspect, and if necessary, repair or replace the detail parts in accordance with the instructions in Table 5.2.2. Inspect the parts in a clean, well-lit room.

Overhaul Kit P/N 212-040-00 is recommended to use for re-assembly of the swing suspension. The overhaul kit contains fasteners, bushings and other recommended items to be replaced. The cargo hook has its own Overhaul Kit, refer to its CMM.

Table 5.2.2 Swing Suspension Inspection Criteria

Seq	Component	Inspection Criteria & Limit	Repair Action	Finish	Recommended replacement at overhaul
1.	Gimbal (1.1), Gimbal (36.1)	Surface cracks – inspect under illuminated magnification (minimum 2X / 4 diopter).	None. Cracks of any size are cause for part replacement.	N/A	No
2.	Frame Strut Weldment (2)	Dents, gouges, and scratches – 0.020 in. (0.51 mm) deep.	Blend at 10:1 ratio as required to provide smooth transitions.	Protect affected surfaces with MIL-PRF-23377 Type 1 primer or equivalent and MIL-PRF-85285 Type 1 coating or equivalent.	No
		Surface cracks – inspect under illuminated magnification (minimum 2X / 4 diopter).	None. Cracks of any size are cause for part replacement.	N/A	
3.	Swing Frame Weldment (3)	Dents, gouges, and scratches – 0.020 in. (0.51 mm) deep.	Blend at 10:1 ratio as required to provide smooth transitions.	Protect affected surfaces with MIL-PRF-23377 Type 1 primer or equivalent and MIL-PRF-85285 Type 1 coating or equivalent.	No
		Surface cracks – inspect under illuminated magnification (minimum 2X / 4 diopter).	None. Cracks of any size are cause for part replacement.	N/A	
		Bent lateral tube, gap measured along a straight edge is greater than .35” (see figure below).	None.	N/A	
					

5.2 5 Year/1000 Hour Inspection continued

Table 5.2.2 Swing Suspension Inspection Criteria continued

Seq	Component	Inspection Criteria & Limit	Repair Action	Finish	Recommended replacement at overhaul
4.	Pivot Shaft (4)	Wear on outside diameter, diameter less than 0.990" (25.1 mm).	None	N/A	No
5.	Shaft Cap (5)	Dents, nicks, cracks, gouges, scratches and corrosion – 0.060 in. (1.52 mm) deep	Blend at 10:1 ratio as required to provide smooth transitions.	Passivate per AMS-QQ-P-35 or ASTM A967.	No
6.	Bumper (6)	Gouges and scratches greater than .060" deep. Splitting.	None.	N/A	No
7.	DU Bearing (1.2, 1.3, 15, 36.2, 36.3, 42.4, 42.5)	Wear – more than 50% copper showing.	None	N/A	Yes
8.	Rod End (13)	Wear on or elongation of inside diameter of spherical bearing, diameter greater than .330 in. (8.4 mm). Binding of spherical bearing in its housing.	None	N/A	No
9.	DU Thrust Washer (15)	Wear on side facing Gimbal – more than 50% copper showing.	None.	N/A	Yes
10.	Load Cell Assembly (16)	Dents, gouges, and scratches greater than .020 in. deep in the load link.	Blend at 10:1 ratio as required to provide smooth transitions.	Passivate per AMS-QQ-P-35 or ASTM A967.	No
		Elongation of or wear on inside diameter of upper lugs, diameter greater than .759 in. (19.3 mm). Galling, pitting on inside diameter.	None.	N/A	
		Dents, gouges, and scratches greater than .060 in. (1.5 mm) deep in the Covers.	Blend at 10:1 ratio as required to provide smooth transitions.	Apply conversion coating (MIL-DTL-5541) to affected Cover surfaces.	
		Wear on ID of bushing - 0.520 in. (13.2 mm)	None.	N/A	
11.	Gimbal Shaft (17)	Wear on OD - .732 in. (18.6 mm) Galling, pitting.	None	N/A	No
12.	Retaining Bushing (18)	Wear on shoulder OD - .420 in. (10.7 mm)	None	N/A	No
13.	Hook Bumper (23)	Gouges and scratches greater than .060 in (1.5 mm) deep.	None	N/A	No

5.2 5 Year/1000 Hour Inspection continued

Table 5.2.2 Swing Suspension Inspection Criteria continued

Seq	Component	Inspection Criteria & Limit	Repair Action	Finish	Recommended replacement at overhaul
14.	Attach Bolt (24)	Wear on OD - .495 in. (12.6 mm)	None	N/A	No
15.	Suspension Cable (37, 38) Fork End Fittings	Wear on ID of clevis holes - 0.397 in. (10.1 mm)	None	N/A	No
		Dents, gouges and scratches – 0.020 in. (0.51 mm) deep	Blend at 10:1 ratio to provide smooth transitions.	Parts are 303 stainless steel, no touch up finish required.	No
		Surface cracks – inspect under illuminated magnification (minimum 2X / 4 diopter).	None	N/A	No
16.	Suspension Cable (37, 38) Wire Rope	Inspect for damage as detailed in 100 Hour/Annual Inspection	None	N/A	No
17.	Quick Release Pin (40)	Wear on OD - 0.362 in. (9.2 mm)	None	N/A	No
18.	Shackle (42.3)	Dents, gouges and scratches – 0.020 in. (0.51 mm) deep	Blend at 10:1 ratio to provide smooth transitions.	Protect affected surfaces with MIL-PRF-23377 Type 1 epoxy primer or equivalent and MIL-PRF-85285 Type 1 coating or equivalent.	No
19.	All remaining nuts, bolts, cotter pins, washers	Wear, corrosion or deterioration.	None	N/A	Yes
20.	S/N Plate	Damaged or illegible	None	N/A	No

5.2 5 Year/1000 Hour Inspection continued

Swing Frame Re-assembly

Re-assemble the suspension frame per the following.

1. Install replacement self-lubricating bushings with wet zinc chromate primer (TTP-1757 or equivalent) applied to the inside diameter of the mating bore.
2. Apply a light film of grease (Mobilgrease 28 is recommended) to the outside diameter of the Pivot Shaft (4) and insert it through one frame half, thrust washer* (15), Gimbal Assembly (1), thrust washer* (15) and through second frame half. Do not fully seat frame halves yet.

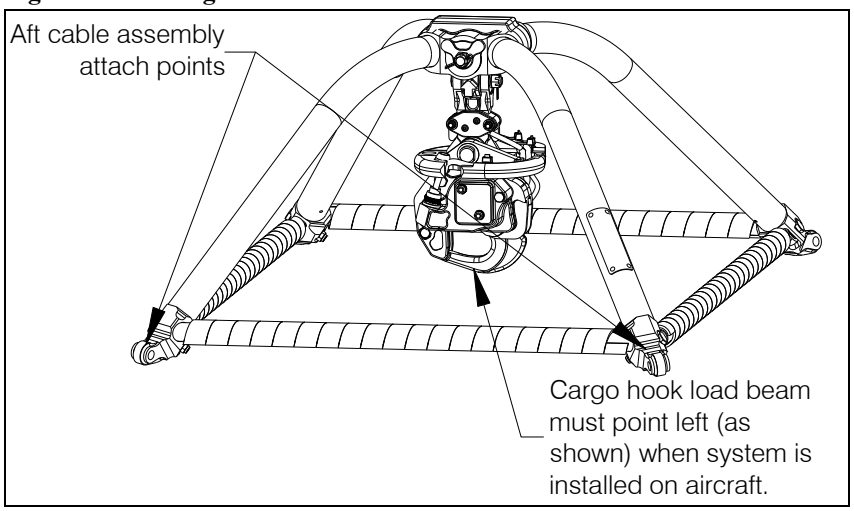
   *Ensure Teflon impregnated wear surfaces (darker sides) of thrust washers are facing gimbal.
3. Rotate the Pivot Shaft so that the raised "keys" at each end are horizontal.
4. Align the rod ends (13) to seat in the pockets of the swing frame feet.
5. Before fully seating and securing the frame halves together position the bumper (6) in between them.
6. Capture each end of the Pivot Shaft with Shaft Caps (5) and insert bolt (10) through, and thread the nut (9) on. Ensure that the rod ends are aligned with the holes in both feet.
7. Torque the nut (9) to 20 ft-lbs. Rotate the nut to the next castellation, not to exceed 30 ft-lbs.
8. Install and secure cotter pin (8).
9. Secure rod ends to frame feet with bolt (12) and nut (7). Torque to 8 – 12 ft-lbs.
10. Ensure the rod ends at each end of the frame struts are parallel, i.e.- the rod ends should be able to be rotated within the limits of the pockets in the frame feet. If necessary, loosen a jam nut (11), rotate the strut so the tightened rod end is against the pocket, rotate other rod end in the same direction (to be parallel), and tighten its jam nut.
11. Attach the Load Cell (16) to the Gimbal Assembly (1) per section 25.17.
12. Attach the Cargo Hook (28) and Hook Bumper (23) to the Load Cell per section 25.17.
13. Apply grease (Mobilgrease 28 is recommended) to the holes of the Fork End Fittings of the Forward and Aft Suspension Cable Assemblies (37, 38).

5.2 5 Year/1000 Hour Inspection continued

14. Assemble the Gimbal Assemblies (36) onto the Fork End Fittings of the suspension cable assemblies with bolt (39), washer (33), and nut (34).
15. Tighten nut (34) to finger tighten then rotate to next castellation if necessary to insert cotter pin. Ensure the Gimbal Assembly pivots freely on the bolt (back off to previous castellation if it does not). Install cotter pin (35).
16. Apply grease (Mobilgrease 28 is recommended) to the shank of the bolts (31).
17. Fasten the Gimbal Assemblies of the two longer Aft Suspension Cable Assemblies onto the aft feet of the Swing Frame Assembly and the two shorter Aft Suspension Cable Assemblies onto the forward feet of the Swing Frame Assembly (reference Figure 5.2.5 for orientation).

    At each joint, assemble with bolt (31), washers (33), Standoff Bushings (32), and nut (34). Tighten nut to 95-110 in-lb and rotate to next castellation to insert cotter pin (35). Ensure each Gimbal rotates freely on the frame foot.

Figure 5.2.5 Swing Frame Orientation



18. Re-install the suspension system on the aircraft per Section 25.17.

5.3 Cargo Hook Overhaul Schedule

Time Between Overhaul (TBO) for the cargo hook: 5 years or 1000 hours of external load operations, whichever comes first.



*Hours of external load operations should be interpreted to be **(1)** anything is attached to the primary cargo hook (whether a useful load is being transported) and **(2)** the aircraft is flying. If these conditions are **NOT** met, time does **NOT** need to be tracked.*

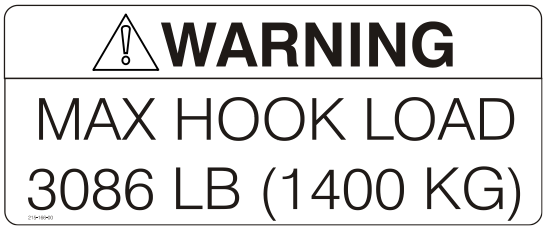
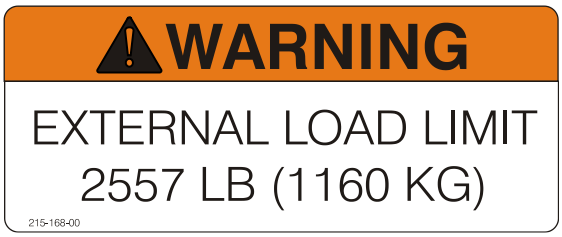
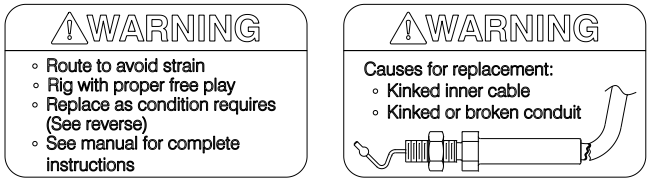
Overhaul instructions for the cargo hook are contained in Component Maintenance Manual 122-017-00. Contact Onboard Systems for guidance to locate authorized overhaul facilities.

Section 11
Placards and Markings

11.1 Placards

The 200-280-04 Cargo Hook Swing Suspension System Kit includes the following placards shown in Table 11.1.

Table 11.1 Cargo Hook Suspension System Placards

Placard part number and appearance	Location
P/N 215-166-00  or P/N 215-168-00  dependent on the model of AS350 on which the system is installed.	Located on the belly of the aircraft near the cargo hook suspension in clear view of the ground support personnel.
P/N 215-271-00  	Located on the swing suspension frame near the serial number plate.
P/N 215-272-00  One Side Opposite Side	Located on the manual release cable, near the cargo hook.

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Section 12
Servicing
12.2 Lubrication Information

Lubrication of the Cargo Hook Swing Suspension system is required every 500 hours of operation. To obtain maximum life under severe duty conditions such as logging or seismic work, it is recommended to lubricate the Swing Suspension every 250 hours. Recommended types of lubricant are AeroShell 17 (MIL-PRF-21164) or Mobilgrease 28 (MIL-PRF-81322).

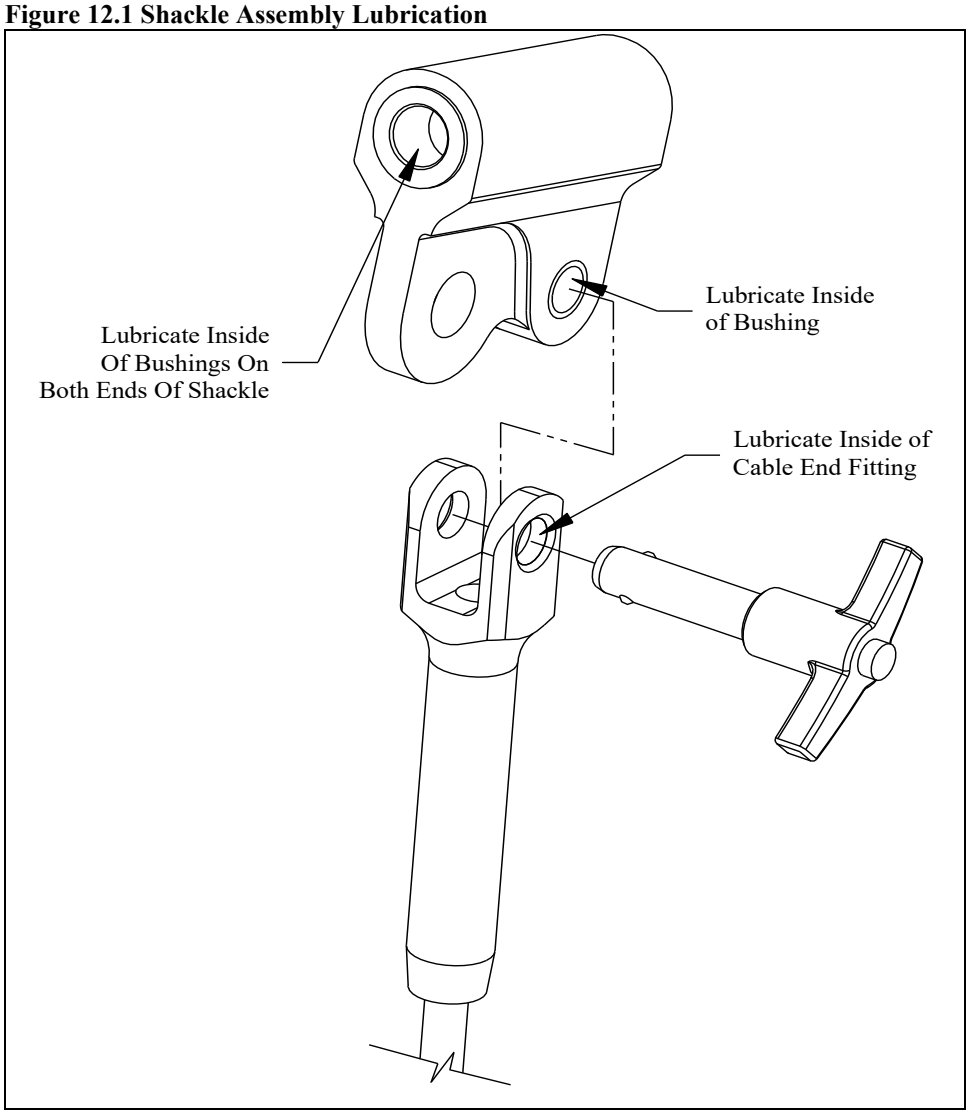
Lubricate the Cargo Hook Swing Suspension at points noted in Figure 12.1 and 12.2.

12.2 Lubrication Information, continued

Shackle Assembly Lubrication

Remove the Shackle Assemblies from the aircraft hard points and lubricate them and the mating fittings on the suspension cables as shown in Figure 12.1. This applies to all four Shackle Assemblies on the helicopter.

Figure 12.1 Shackle Assembly Lubrication

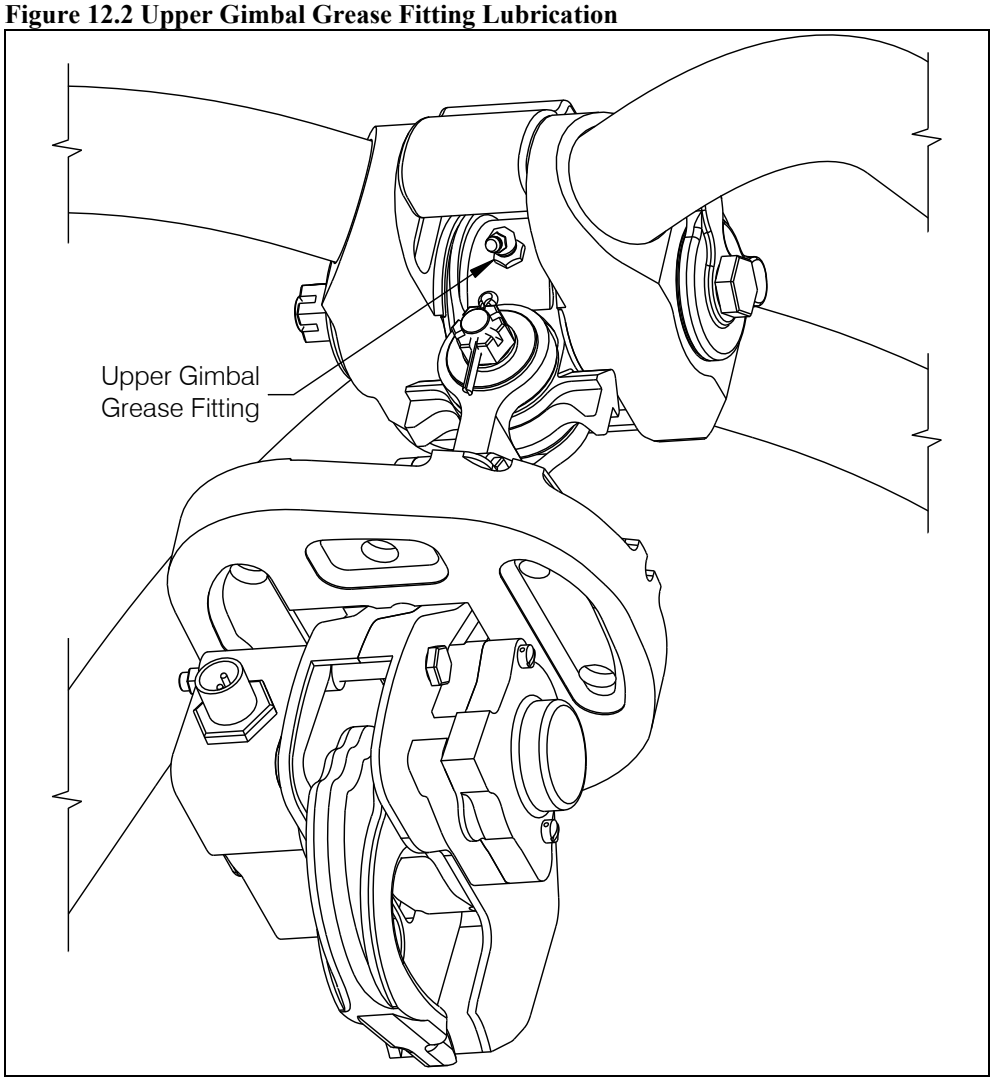


12.2 Lubrication Information, continued

Upper Gimbal Grease Fitting

Lubricate Upper Gimbal Assembly at the grease fitting located as shown in Figure 12.2. You may have to rotate the hook slightly to access the grease fitting.

Figure 12.2 Upper Gimbal Grease Fitting Lubrication



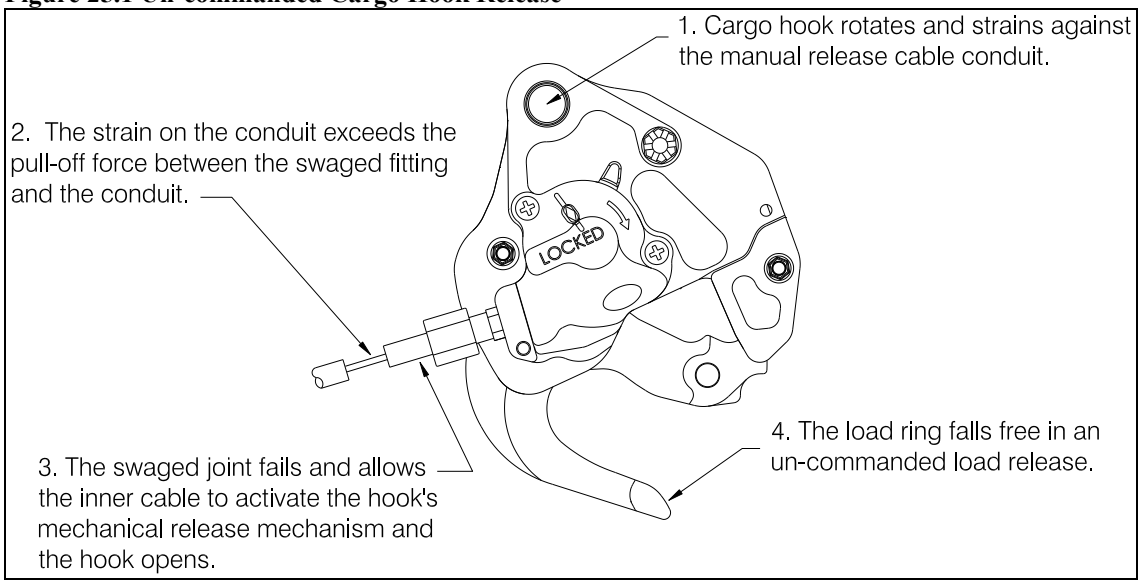
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Section 25
Equipment and Furnishings

WARNING

Un-commanded cargo hook release will happen if the manual release cable is improperly restrained. The cable must not be the stops that prevent the Cargo Hook from swinging freely in all directions. If the Cargo Hook loads cause the hook to strain against the manual release cable the swaged end of the cable may separate allowing the inner cable to activate the cargo hook manual release mechanism. The result is an un-commanded release. Ensure that no combination swing frame or Cargo Hook position is restrained by the manual release cable.

Figure 25.1 Un-commanded Cargo Hook Release



25.1 Cargo Hook Connector

Listed below is the pin out for the cargo hook connector.

Table 25.1 Cargo Hook Connector

Pin	*Function*
A	Ground
B	Positive

25.2 Description

The Cargo Hook Swing Suspension System consists of four primary subsystems, these are the Swing Suspension Assembly, Manual Release System, Electrical Release System, and Load Weighing System.

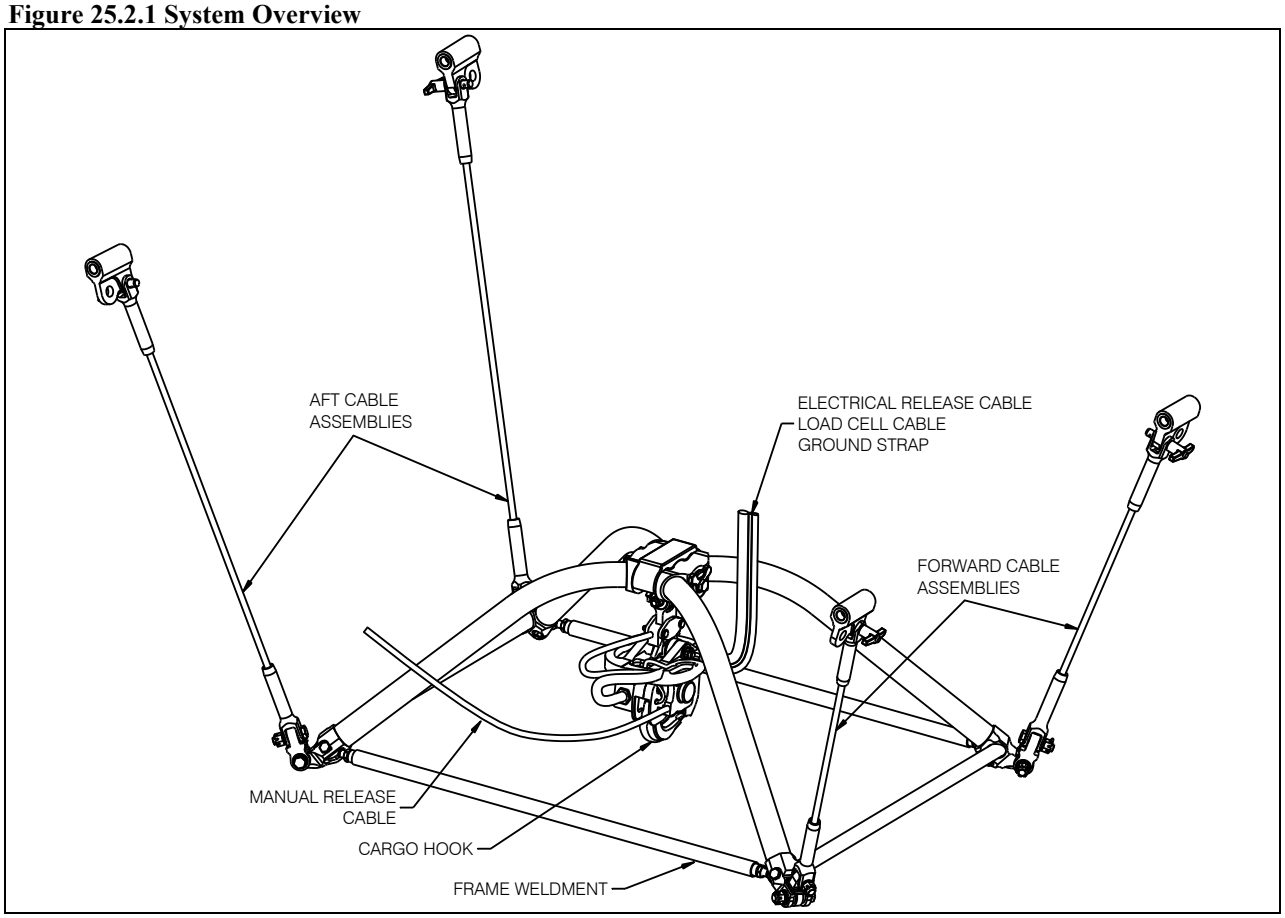
The Swing Suspension Assembly is attached to hard points and suspended below the belly of the helicopter by its four cable assemblies. The cable assemblies are attached to a frame assembly, which supports the cargo hook and a load cell through a gimbal (ref. Figure 25.2.1).

The Electrical Release System provides a means to release a cargo hook load using a switch in the cockpit.

The Manual Release System provides an additional means to release a cargo hook load and consists of a release lever mounted to the collective that actuates a cable that is routed to the cargo hook.

The Load Weighing System consists of an indicator mounted within the cockpit, the load cell on the suspension, and associated wiring.

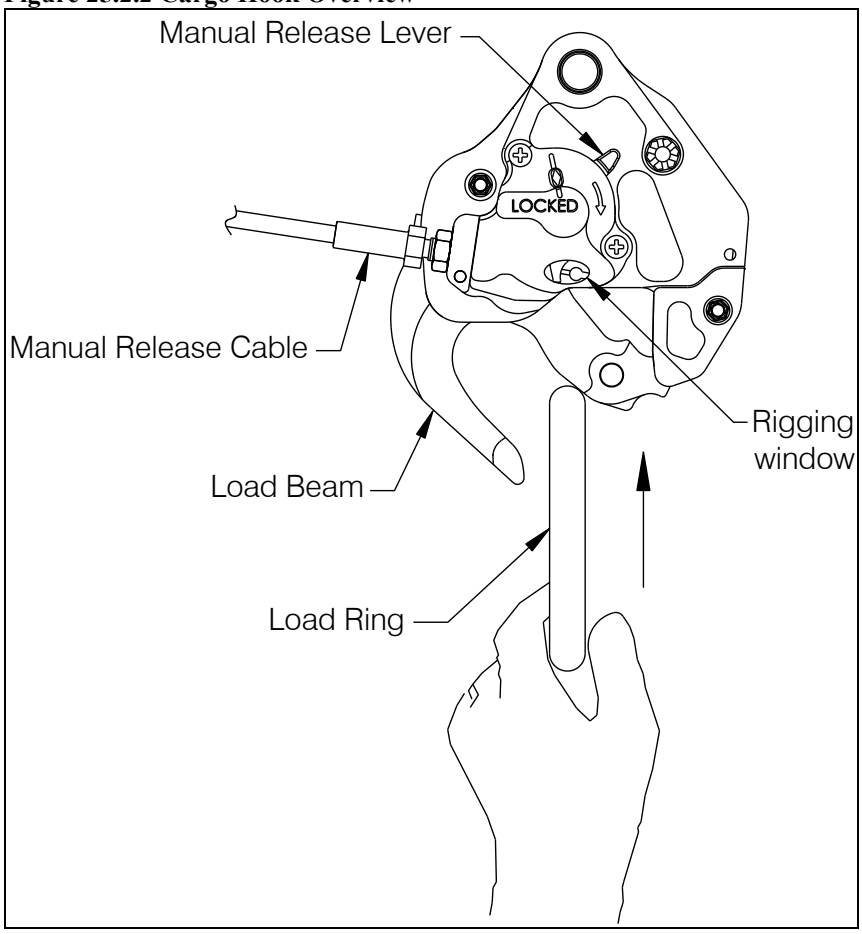
Figure 25.2.1 System Overview



25.2 Description continued

The cargo hook is the means used to attach an external load to the aircraft. A load is attached to the cargo hook by passing a load ring into the throat of the load beam and pushing the ring against the upper portion of the load beam throat (see Figure 25.2.2), which will initiate the hook to close. In the closed position, a latch engages the load beam and latches it in this position. A load release can be initiated by three different methods. Normal release is achieved by pilot actuation of a push-button switch in the cockpit. When the push-button switch is pressed, it energizes the solenoid in the cargo hook, and the solenoid opens the latch in the internal mechanism. In the event of an electrical failure, load release can be achieved by operating the manual release cable. The release cable actuates the internal mechanism of the cargo hook to unlatch the load beam. A rigging window provides a means to verify the manual release cable setting with respect to the internal mechanism. Ground personnel can also release the load by actuating a manual release lever located on the side of the cargo hook (see Figure 25.2.2).

Figure 25.2.2 Cargo Hook Overview

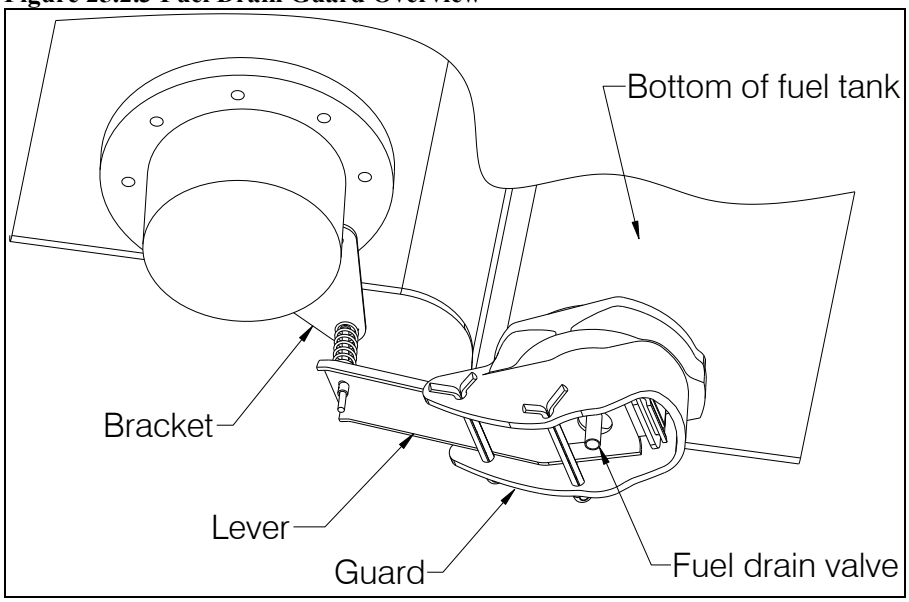


25.2 Description continued

The fuel drain guard protects the fuel drain valve on AS350 helicopters from accidentally being opened. The fuel drain valve is located on the bottom of the fuel tank and extends below the belly of the helicopter. This location makes it vulnerable to damage or un-commanded fuel drainage on helicopters equipped with a cargo hook swing suspension. The most common occurrence of the cargo hook swing suspension striking the fuel drain valve happens when the helicopter lands on snow or on uneven terrain. The swing suspension has limited ground clearance and when the skid gear sinks into the snow, the swing suspension is pushed upward into the fuel drain valve, opening it and causing fuel to drain. The fuel drain valve can also be opened in flight by the swing suspension flying vertically due to aerodynamics when ferrying with no load or from recoil effects from releasing large cargo hook loads.

The Fuel Drain Guard interfaces with the Airbus Helicopters' existing valve, lever and control cable. The kit includes a Bracket, which replaces Airbus Helicopters' bracket and provides an optimized mounting point for the control cable.

Figure 25.2.3 Fuel Drain Guard Overview



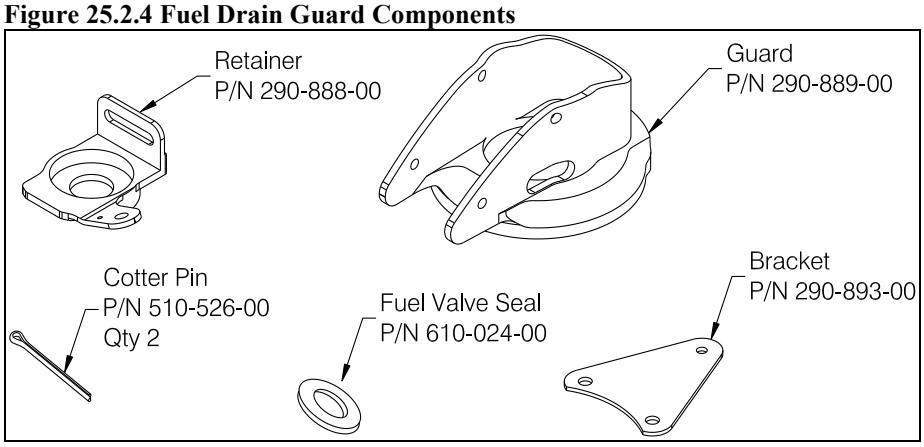
NOTICE

The fuel drain guard was not included with earlier kits and is not installed with the Airbus crash resistant fuel tank so may not be present.

25.2 Description continued

Figure 25.2.4 shows the components of the Fuel Drain Guard installation.

Figure 25.2.4 Fuel Drain Guard Components



25.5 Component Weights

The weights and cgs of the systems are listed in Table 25.2.

Table 25.2 Component Weights and CGs

Item	Weight	Station
Removable Provisions*	30.0 lbs (13.6 kg)	133 in (3375 mm)
Fixed Provisions**	5.5 lbs (2.5 kg)	110 in (2794 mm)
Fuel Drain Guard***	0.40 lbs (.18 kg)	135 in. (3430 mm)
Total	35.9 lbs (16.3 kg)	129.4 in (3288 mm)

* The removable provisions include the swing suspension w/ hook, external manual release cable, and external electrical release cable. These items are easily removed if they are not needed on the helicopter's mission. Refer to Suspension System Removal in Section 25.16 for removal instructions.

** The fixed provisions are those items of the kit that remain on the aircraft. These include the fixed manual release cable, internal electrical wire harnesses, the load weigh indicator, and the miscellaneous brackets that support these items.

*** An optional item on earlier kits and not installed with the Airbus crash resistant fuel tank.

25.12 Storage Instructions

Clean the exterior Cargo Hook and suspension components thoroughly of excess dirt and grease with a rag before packaging. Pack the unit in a heat-sealable package. If the unit is to be stored for long periods in a tropical climate it should be packed in a reliable manner to suit local conditions. Refer to MIL-PRF-23199 and MIL-STD-2073-1 for additional guidance and to CMM 122-017-00 for storage instructions for the Cargo Hook.

Package the unit in a suitable fiberboard box and cushion the unit to prevent shifting. Seal the fiberboard box with tape and mark the box with the contents and date of packaging.

25.15 Troubleshooting

Table 25.15.1 is provided with the intention of isolating the cause of malfunctions within the system. Sections 25.16 and 25.17 include instructions for removing and replacing defective components. Refer to the appropriate Airbus Helicopters maintenance manuals for guidance on procedures relating to Airbus Helicopters parts that interface with this suspension system.

Table 25.15.1 Troubleshooting

MALFUNCTION	PROBABLE CAUSE	CORRECTIVE ACTION
Cargo hook does not operate electrically or manually.	Defective internal mechanism.	Remove and replace cargo hook (see sections 25.16 and 25.17) or repair per CMM 122-017-00.
Cargo hook does not operate electrically, manual cable release operates normally.	Open electrical circuit, faulty wiring, fuse, switch or solenoid.	Disconnect cable from electrical connector on cargo hook. Using multi-meter, check for 3.0 to 4.0 ohms between pins A and B of electrical connector (see note 1 below). If open indication is obtained, remove and replace cargo hook (see sections 25.16 and 25.17) or repair per CMM 122-017-00.
Cargo hook operates electrically, but not manually.	Defective manual release cable. Defective manual release system.	Inspect manual release cable and cable connection to Cargo Hook. Remove and replace cargo hook (see Sections 25.16 and 25.17) or repair per CMM 122-017-00.
Load beam fails to re-latch after being reset.	Defective latch mechanism.	Remove and replace cargo hook (see sections 25.16 and 25.17) or repair per CMM 122-017-00.
Force required to release hook with lever on collective exceeds 14 lbs.	High cable friction or friction in internal mechanism of hook.	Remove cable from hook and check cable and hook independently (see below) to determine cause.
With release cable disconnected at hook, the force required to move manual release lever on collective exceeds 6 lbs.	Kinks or wear in cable, frozen water in cable, debris or damage to cable quick disconnect fitting or lever mechanism on cyclic	Inspect individual components to isolate problem. Remove and replace defective parts (see Sections 25.16 and 25.17 for remove and replace instructions for manual release cable).
Cargo hook manual release cable pull-off force exceeds 8 Lbs. (at the hook).	Friction in internal mechanism.	Remove and replace cargo hook (see Section 25.16 and 25.17) or repair per CMM 122-017-00.
Cargo hook fails to open or re-lock properly.	Failure to open or re-lock properly.	Remove and replace cargo hook or repair per CMM 122-017-00 (see Sections 25.16 and 25.17).
Fuse/circuit breaker opens when cargo hook is energized.	Short in the system, faulty wiring, fuse/circuit breaker or solenoid.	Check for shorts to ground along length of wire harness (see note 2). Check solenoid resistance (see note 1), repair or replace defective parts.

Table 25.15.1 Troubleshooting continued

Load Weigh Indicator does not light up.	Faulty wiring, fuse, or circuit breaker.	Check the fuse/circuit breaker (refer to Airbus Helicopters maintenance manuals) and wiring (see Note 2). If this doesn't help, remove and replace indicator per sections 25.16 and 25.17.
The displayed load on the Load Weigh Indicator is incorrect.	Incorrect calibration code.	Ensure the correct calibration code has been entered (see Note 3).
Indicator displayed load is not stable.	Dampening level is too low.	Adjust the dampening level to a higher number (see Note 4).
Indicator displayed load takes too long to change the reading when the load is changed.	Dampening level is too high.	Adjust the dampening level to a lower number (see Note 4).
Indicator does not change with changing hook loads.	Defective load cell, indicator failure or damaged wire harness.	Check for damaged wire harness (see note 2), remove and replace wire harness assembly or load cell (see sections 25.16 and 25.17).

Notes:

1. **Checking resistance at pins A and B.**

   Check for 3.0 to 4.0 ohms between pins A and B of electrical connector located on the cargo hook (see below).

Figure 25.15.1 Cargo Hook Electrical Connector

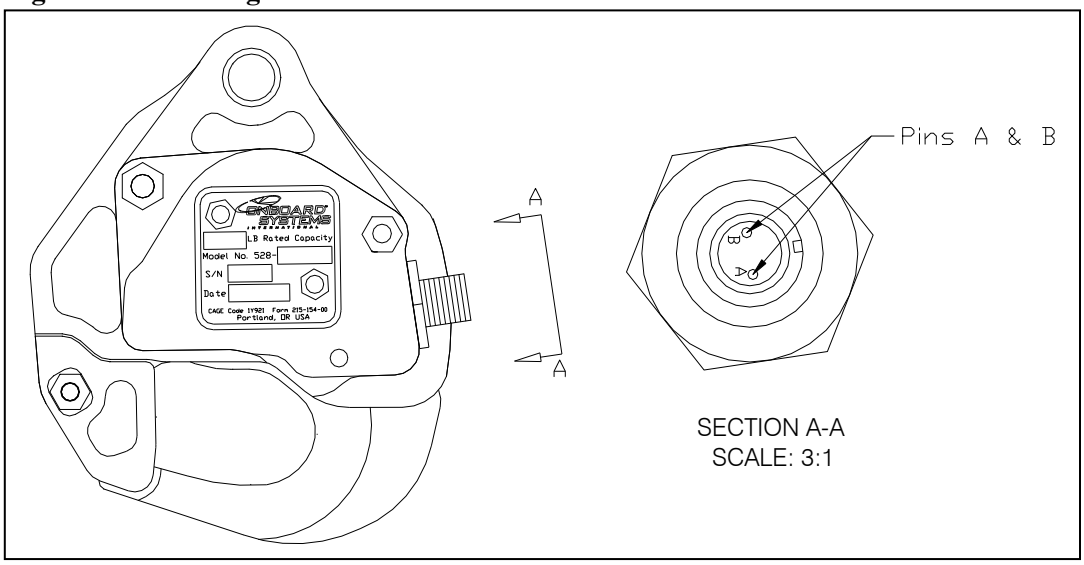


Table 25.15.1 Notes continued:

2. Checking Wire Harnesses.

As appropriate, before working on a circuit, e.g. - inspection, removal-installation of components, check that the aircraft system is not energized:

- "EXT. PWR. BAT." push-button is released.
- External power connector is not supplied
- Further precaution: remove the fuse(s)/circuit breakers from the corresponding circuits (refer to Figure 25.15.3)

The wire harnesses are routed with and secured to existing wire bundles and are located approximately as shown below. Remove lower fairings to inspect wiring underneath the cabin floor. Inspect for general condition and chafing along length of wire runs. See Figure 25.15.3 for electrical schematic.

Figure 25.15.2 Wire Harness Routing

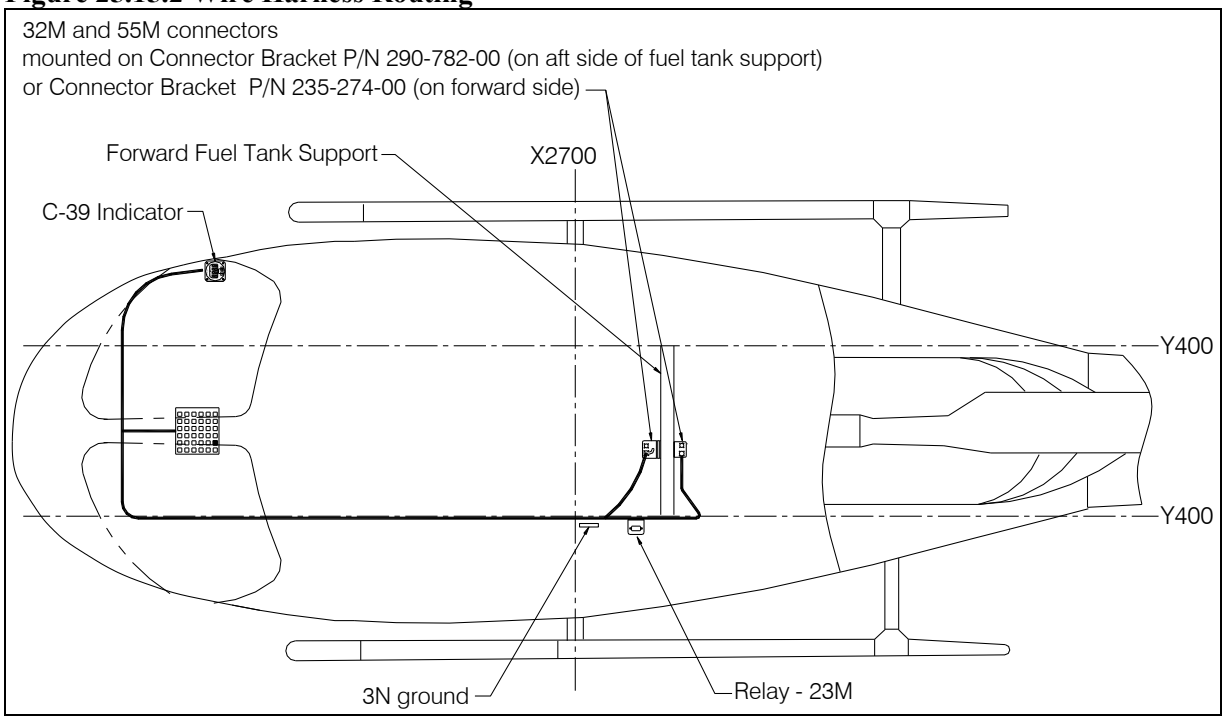


Table 25.15.1 Notes continued:

2. Checking Wire Harnesses continued

The electrical schematic for the electrical release system and the load weigh system is shown below. Airbus Helicopters modification #'s 07-4280 and 07-3450 are reflected below. Earlier Airbus Helicopters configurations which affected how and where wires ME1E, ME2E and ME10E of the electrical release harness and load weigh harness interface with the helicopter are shown on the following page. Refer to the applicable Airbus Helicopters Wiring Diagrams Manual for additional information.

Figure 25.15.3 Electrical Schematic

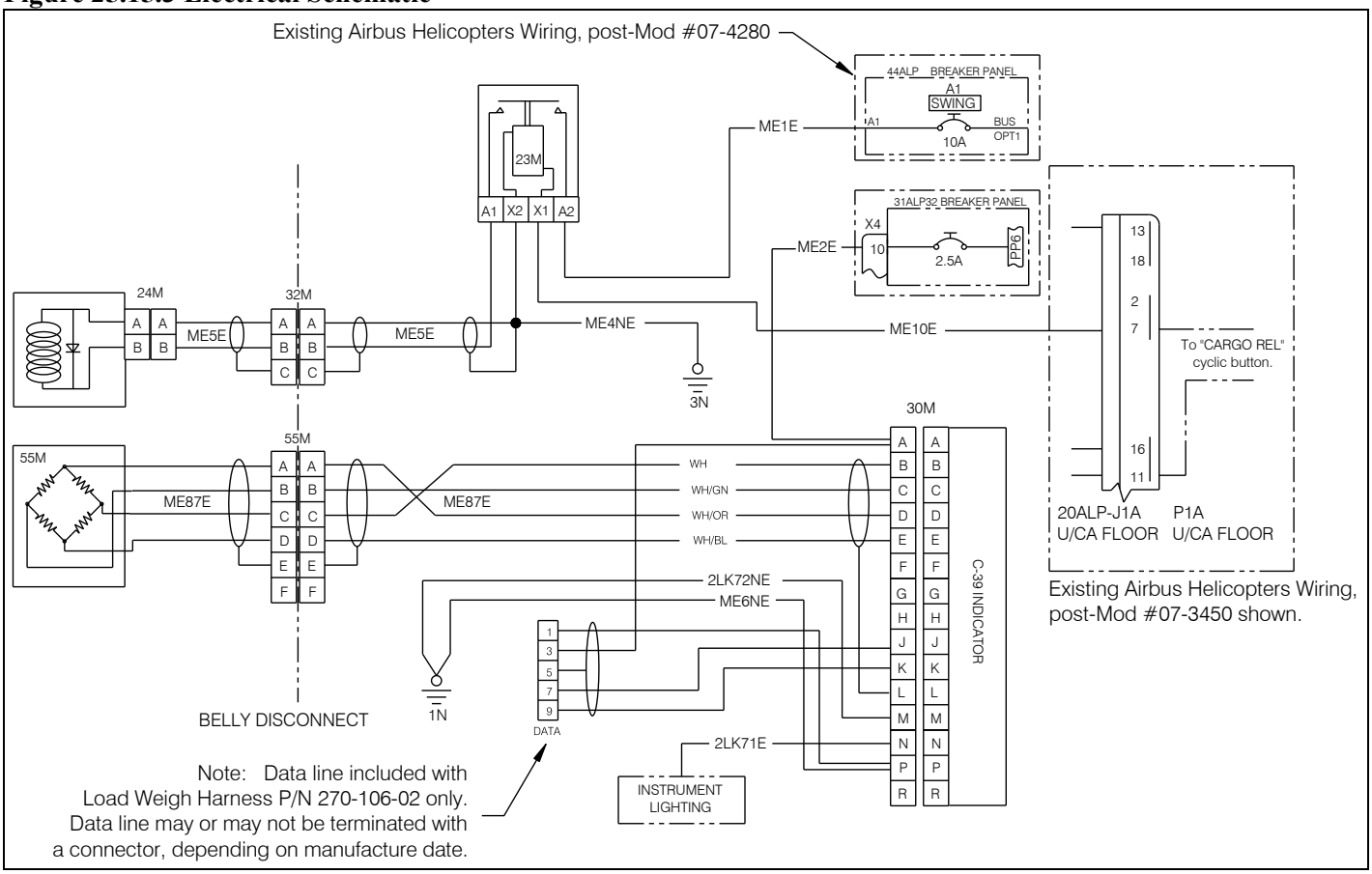
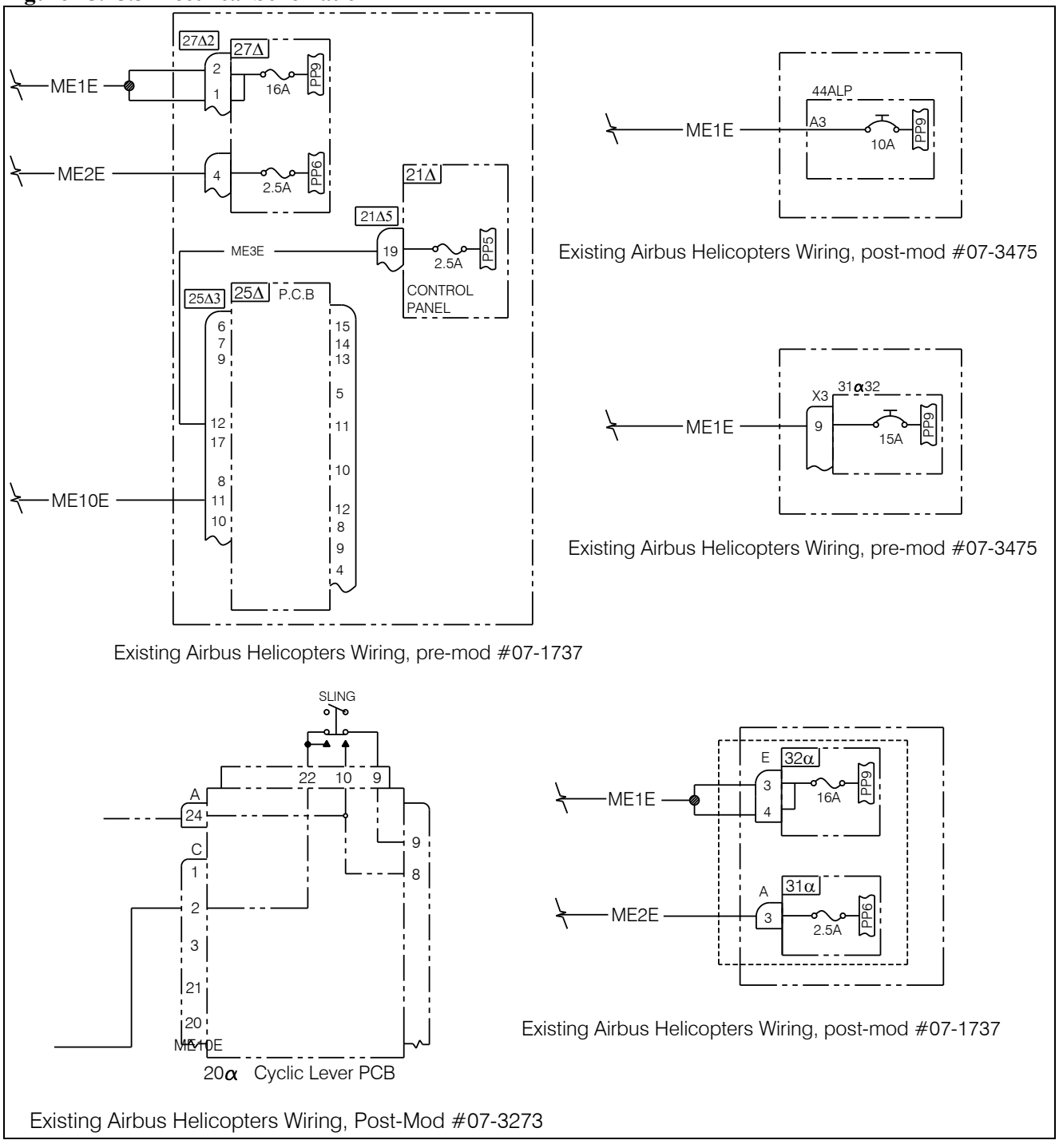


Table 25.15.1 Notes continued:

2. Checking Wire Harnesses continued

Figure 25.15.3 Electrical Schematic continued



Existing Airbus Helicopters Wiring, post-mod #07-3475

Existing Airbus Helicopters Wiring, pre-mod #07-3475

Existing Airbus Helicopters Wiring, pre-mod #07-1737

Existing Airbus Helicopters Wiring, post-mod #07-1737

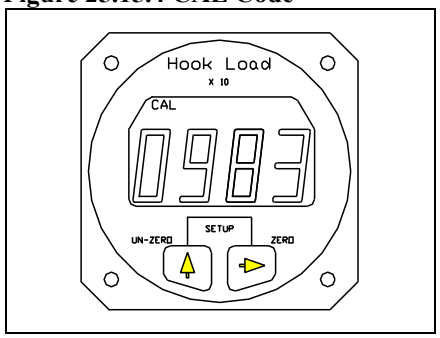
Existing Airbus Helicopters Wiring, Post-Mod #07-3273

Table 25.15.1 Notes continued:

3. Checking Load Weigh Indicator calibration code:

With the Indicator powered up and in the Run Mode, press both buttons at the same time to go to Setup. Scroll through the menu until the word CODE is displayed, then press the Right button. The display should look like this:

Figure 25.15.4 CAL Code

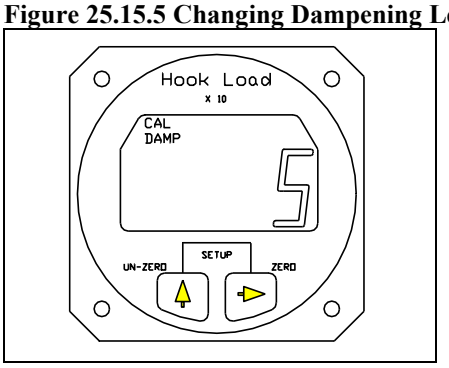


This code should match the code printed on the tag attached to the load cell cable. If this code does not match, contact Onboard Systems for further guidance.

4. Adjusting dampening level:

With the Indicator powered up and in the Run Mode, press both buttons at the same time to go to Setup. Scroll through the menu, using the Left button, until the word DAMP is displayed. To look at or change the Dampening Level press the Right button. The display should look like this:

Figure 25.15.5 Changing Dampening Level



The CAL and the DAMP legend is turned on and the previously set dampening level is displayed. To return to Run without changing the current dampening level press both the Right and Left buttons at the same time. To change the dampening number, use the Left button to scroll the blinking digit to the desired number. Ten dampening levels are available, from 0 through 9. At level 0 the display responds to the slightest change in weight. However, if the load bounced even slightly, the display digits would respond instantly, making the display look unstable. With a dampening level of 9, the display would be stable under the most turbulent conditions, however, it would take several seconds for the display to respond to a change in weight. The ideal dampening level will depend on the flying conditions. A mid range setting of 5 or 6 is usually adequate. After the selection has been made press both the Right and Left buttons at the same time to return to Run.

25.16 Component Removal

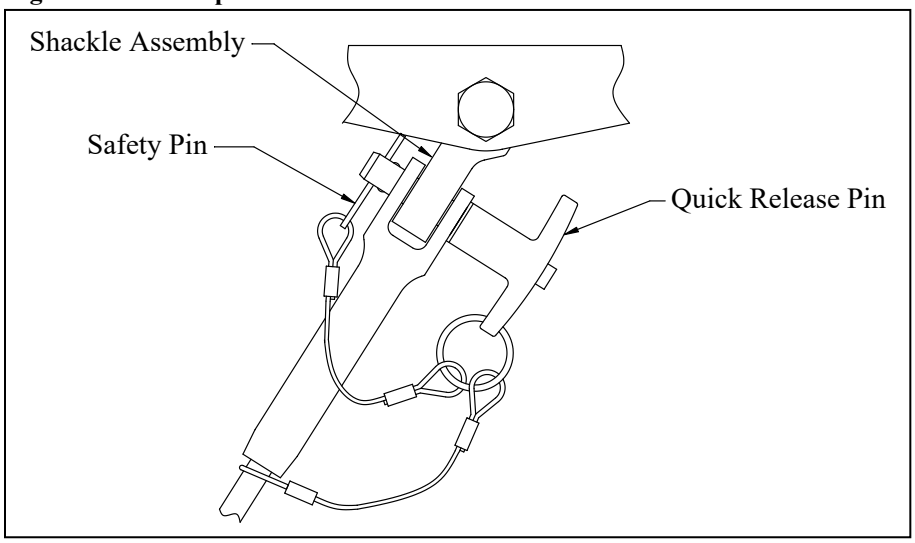
Cargo Hook Removal

1. Remove manual release cover by removing two screws.
2. Disconnect the manual release cable and electrical release harness from the Cargo Hook.
3. Remove the cotter pin (P/N 510-178-00) from the Attach Bolt (P/N 290-775-00).
4. Remove the castellated nut (P/N 510-170-00) from the Attach Bolt.
5. Remove Attach Bolt and all washers.
6. Remove the Cargo Hook from suspension system.
7. Remove the Hook Bumper (P/N 290-774-00) and ground strap from the cargo hook.

Suspension System Removal

1. Disconnect the load cell harness at the bulkhead connector on the belly of the aircraft.
2. Disconnect the electrical release harness at the bulkhead connector on the belly of the aircraft.
3. Disconnect the ground strap at the quick disconnect near the bulkhead connectors on the belly of the aircraft.
4. Disconnect the manual release cable at the quick release fitting.
5. Remove the Safety Pins (P/N 510-464-00) and Quick Release Pins (P/N 290-851-00) at the 4 joints where the suspension cable ends mate with the Shackle Assemblies (P/N 232-137-01) as illustrated below.

Figure 25.16.1 Suspension Attachment Hardware

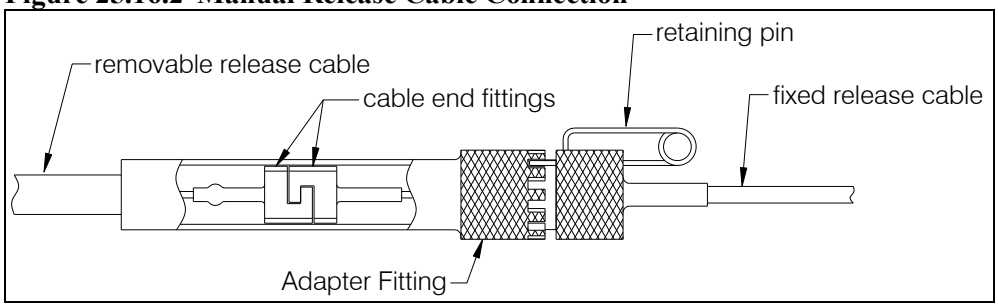


25.16 Component Removal continued

External Manual Release Cable Removal

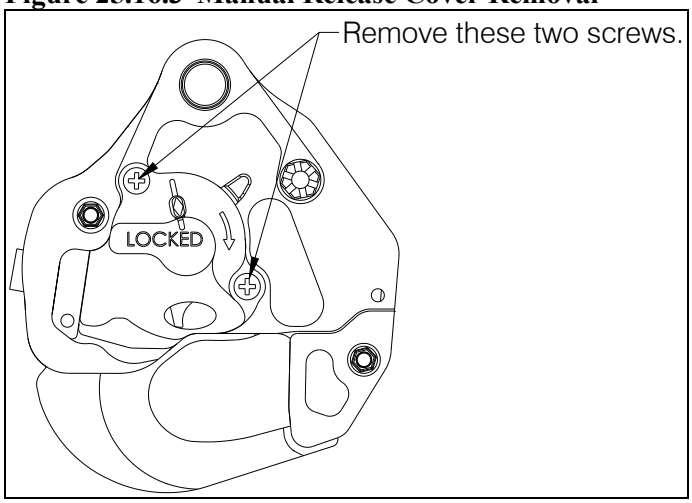
1. Disconnect the cable at the joint with the fixed manual release cable on the belly of the helicopter by disengaging the retaining pin and unthreading the adapter fitting to expose and disengage the cable end fittings.

Figure 25.16.2 Manual Release Cable Connection



2. At the other end of the cable (at the cargo hook) remove the two screws that secure the manual release cover to the hook (see below) and unhook the cable ball end from the fork fitting.

Figure 25.16.3 Manual Release Cover Removal



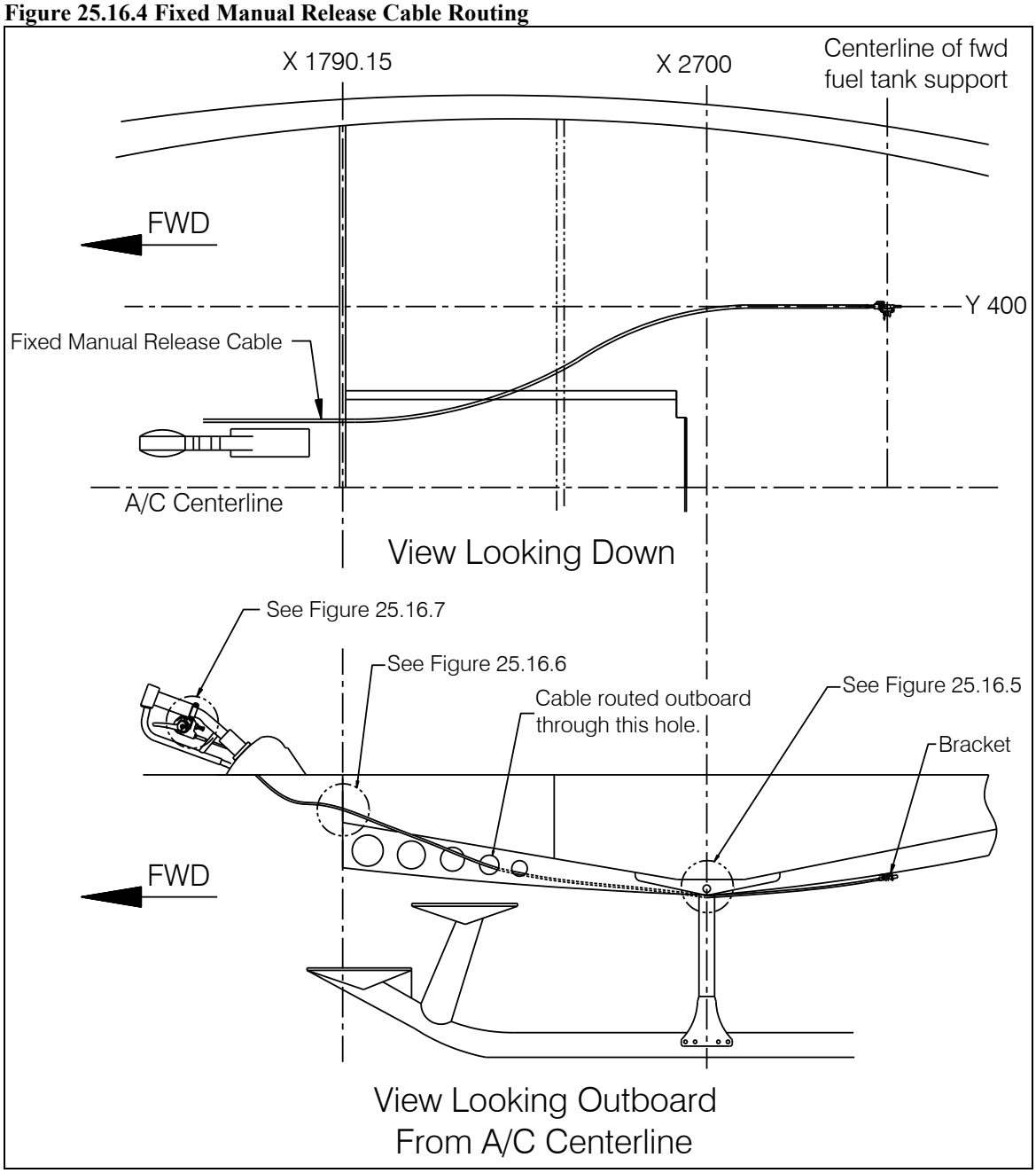
3. Loosen the jam nut and unthread the release cable from the hook.

25.16 Component Removal continued

Fixed Manual Release Cable Assembly Removal

The fixed manual release cable is routed from the release lever mounted to the collective stick to the bracket on the belly of the helicopter where it is mated with the removable section of the cable.

Figure 25.16.4 Fixed Manual Release Cable Routing

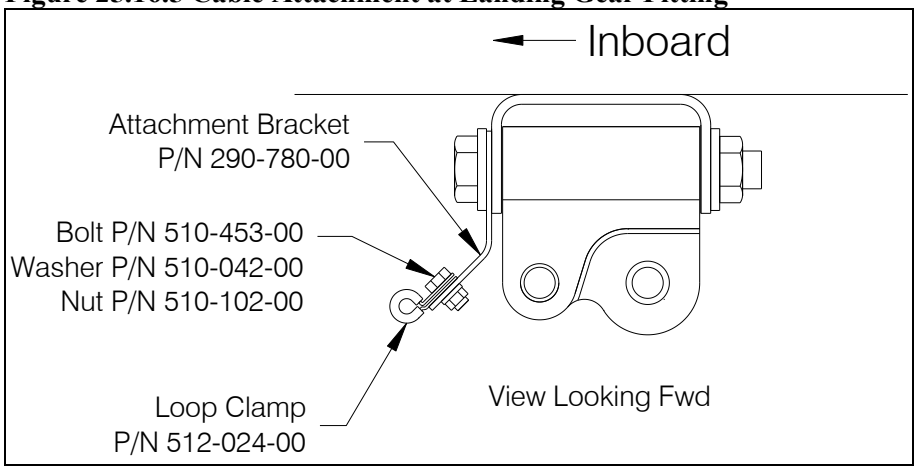


25.16 Component Removal continued

Fixed Manual Release Cable Assembly Removal continued

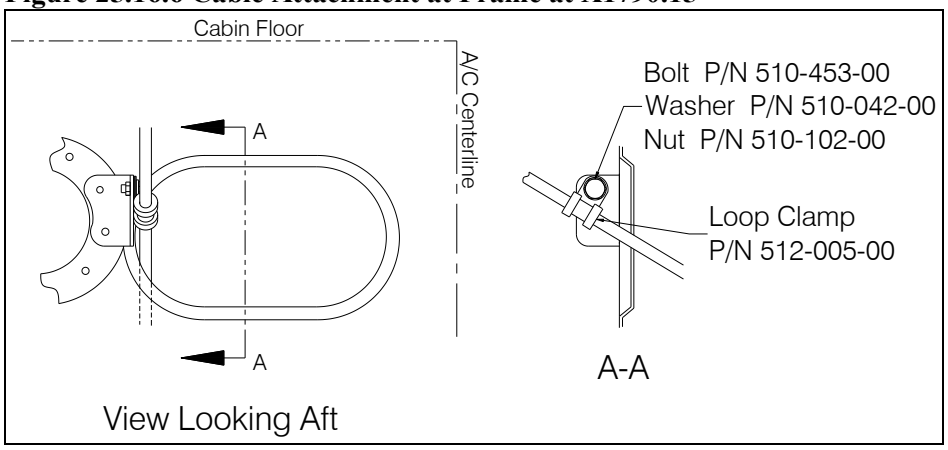
1. Unclip the end of the cable assembly from the bracket on the belly of the helicopter.
2. Remove the cushioned loop clamp from the Attachment Bracket at the RH forward landing gear fitting and remove it from the cable.

Figure 25.16.5 Cable Attachment at Landing Gear Fitting



3. Moving farther forward, remove the loop clamp at the bracket at frame at 1790.15 and remove it from the cable.

Figure 25.16.6 Cable Attachment at Frame at X1790.15

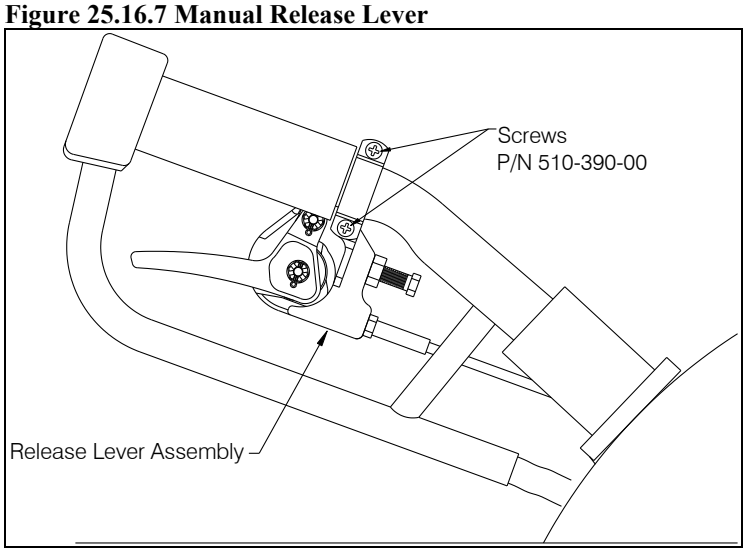


25.16 Component Removal continued

Fixed Manual Release Cable Assembly Removal continued

4. Above the floor and on the collective remove the release lever by removing two screws (see below).

Figure 25.16.7 Manual Release Lever



5. Feed the cable forward and then up through the slot in the floor.

NOTICE

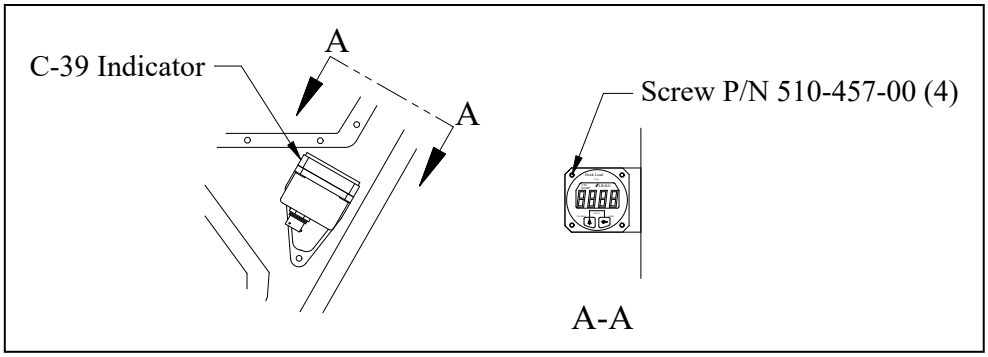
Remove the grommet from the slot to allow the end fitting on the cable to be fed through.

Load Weigh Indicator Removal

The load weigh indicator is located on the RH forward door pillar.

1. Disconnect electrical connector from the back of indicator.
2. Remove the four screws (P/N 510-457-00) that secure the indicator to the mounting bracket and remove the indicator.

Figure 25.16.8 Load Weigh Indicator

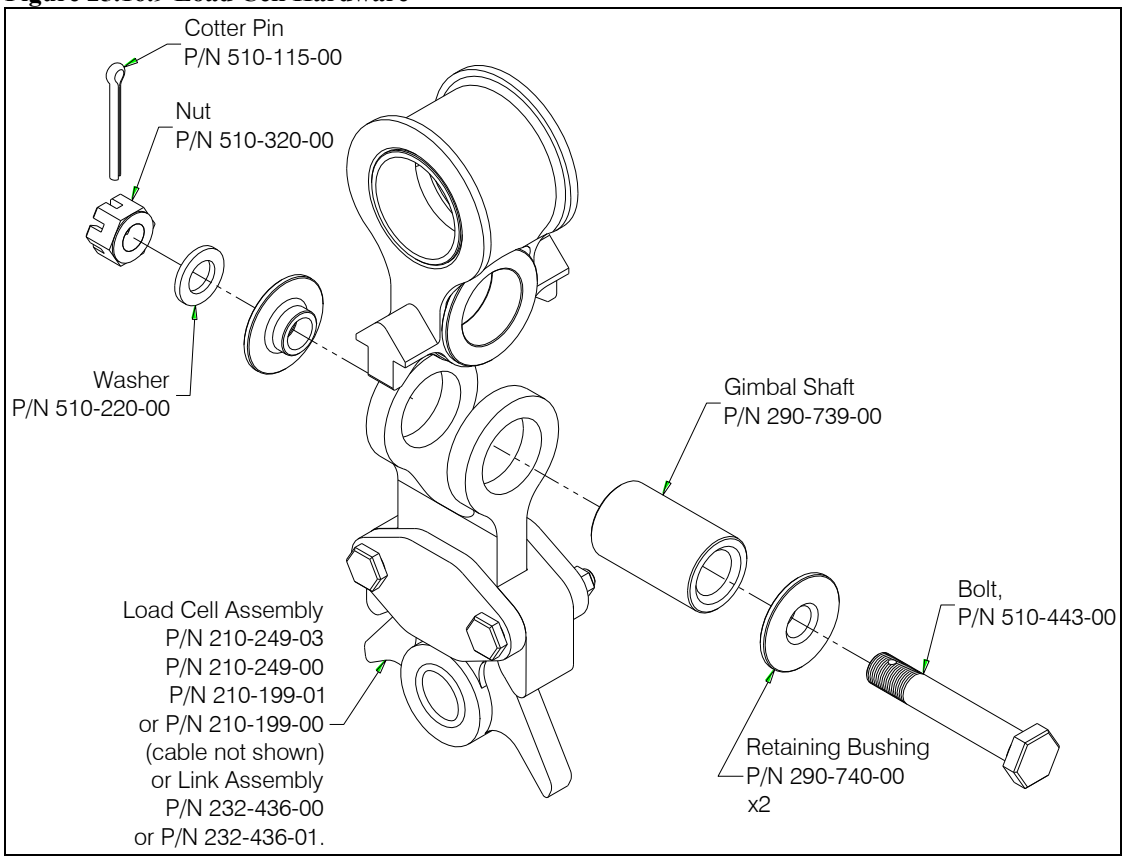


25.16 Component Removal continued

Load Cell Removal

1. Disconnect the electrical connector on the belly of the helicopter.
2. Remove the Cargo Hook per the above instructions.
3. Remove the Load Cell Assembly from the gimbal on the suspension frame (not shown) by removing the hardware as illustrated below.

Figure 25.16.9 Load Cell Hardware



Self Lubricated Bushing Removal

All self-lubricated bushings use an interference fit to hold them in place. Use an arbor press or similar to press the bushings out of bore they are mounted in.

CAUTION

Do not use heat on any of the parts when removing the self-lubricated bushings. These parts are all heat-treated and using heat may affect their mechanical properties.

25.16 Component Removal continued

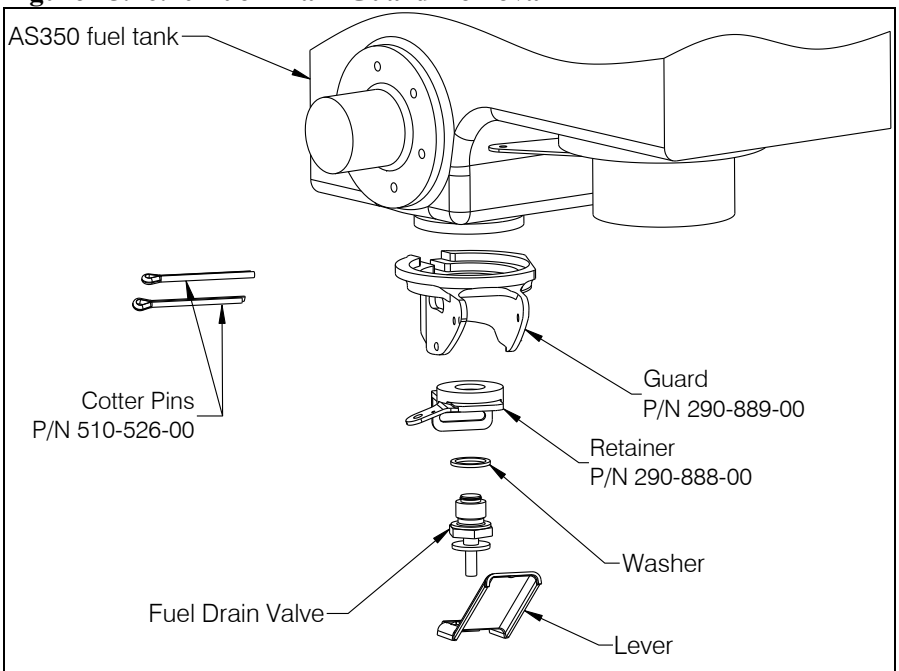
Fuel Drain Guard Removal

NOTICE

The fuel drain guard may not be present with all cargo swing installations (it was not included with earlier kits (prior to 2010)) and is not installed with the Airbus crash resistant fuel tank).

1. Drain fuel tank.
2. Remove cotter pins from Guard.
3. Remove lever.
4. Remove safety wire between fuel tank valve and retainer tab.
5. Remove ground strap from Retainer.
6. Unthread fuel drain valve from tank and remove it and washer.
7. Remove Guard and Retainer.
8. Remove Bracket.

Figure 25.16.10 Fuel Drain Guard Removal



Note: Fuel Drain Valve, Washer, and Lever are Eurocopter parts. Refer to your helicopter's maintenance manual for information regarding these items.

25.17 Component Re-installation

Suspension Re-installation

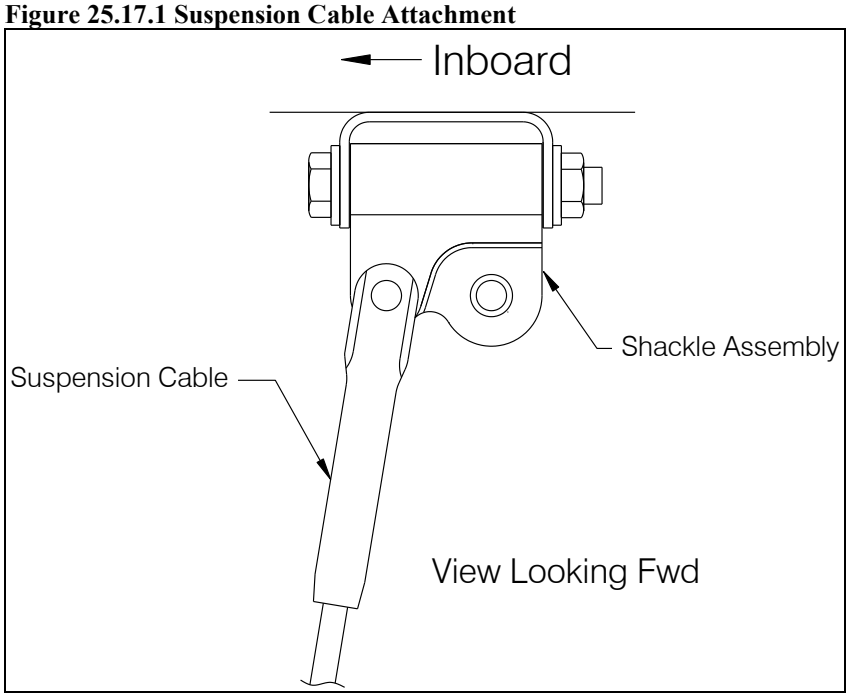
1. Install the Suspension by securing the shorter suspension cables' Clevis Cable Ends (see figure below) to the forward Shackle Assemblies and the longer suspension cables' Clevis Cable Ends to the aft Shackle Assemblies with the quick release pins. Secure quick release pins with attached safety pins.

CAUTION

*Cable Clevis Ends must be pinned to the **inboard** holes of the Shackle Assemblies (as illustrated below).*

2. Connect the load cell cable connector at the bulkhead connector at the belly of the aircraft.
3. Connect the electrical release cable at the bulkhead connector at the belly of the aircraft.
4. Connect the ground strap attached to the cargo hook to the ground strap attached to the aircraft.
5. Re-install the manual release cable per instructions in this section.

Figure 25.17.1 Suspension Cable Attachment

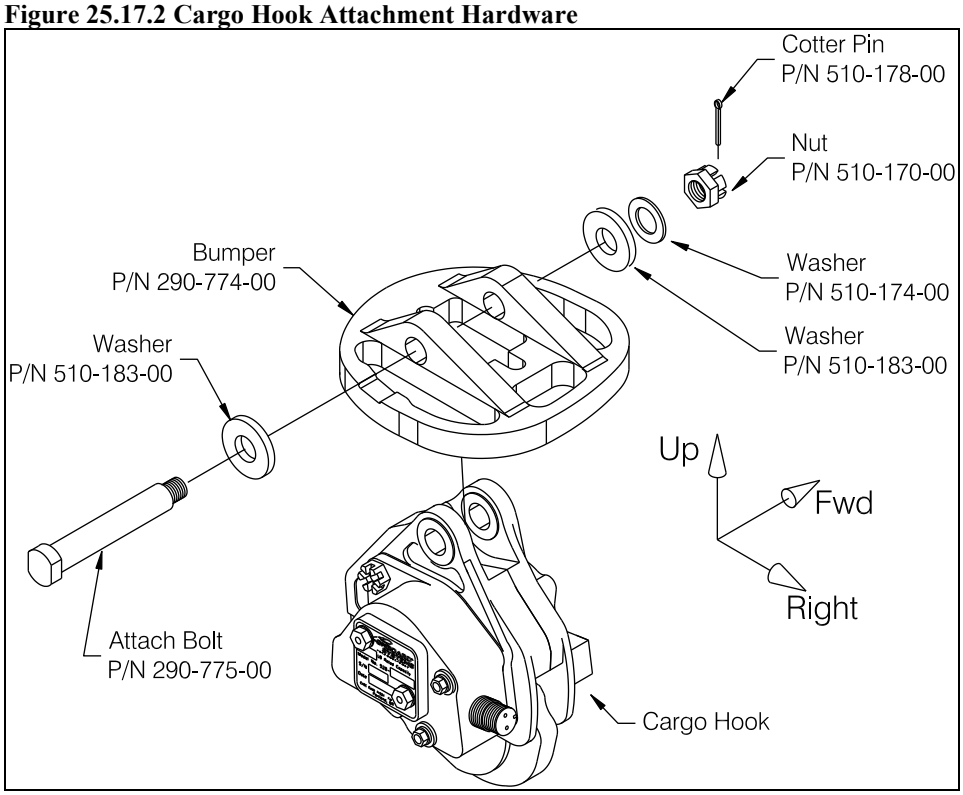


25.17 Component Re-installation continued

Cargo Hook Re-installation

1. Attach the Cargo Hook, P/N 528-029-00 to the suspension system by installing the Bumper P/N 290-774-00 over the Cargo Hook.
2. Install the attach bolt P/N 290-775-00 and washer P/N 510-183-00 as illustrated in Figure 25.17.2
3. Install washer P/N 510-183-00 and washer P/N 510-174-00 over bolt end.
4. Tighten nut P/N 510-170-00 on attach bolt to finger tight, then rotate to next castellation to install and secure cotter pin P/N 510-178-00.

Figure 25.17.2 Cargo Hook Attachment Hardware



CAUTION

The Cargo Hook Load Beam must point to the left side of the helicopter when looking from the rear (as shown above).

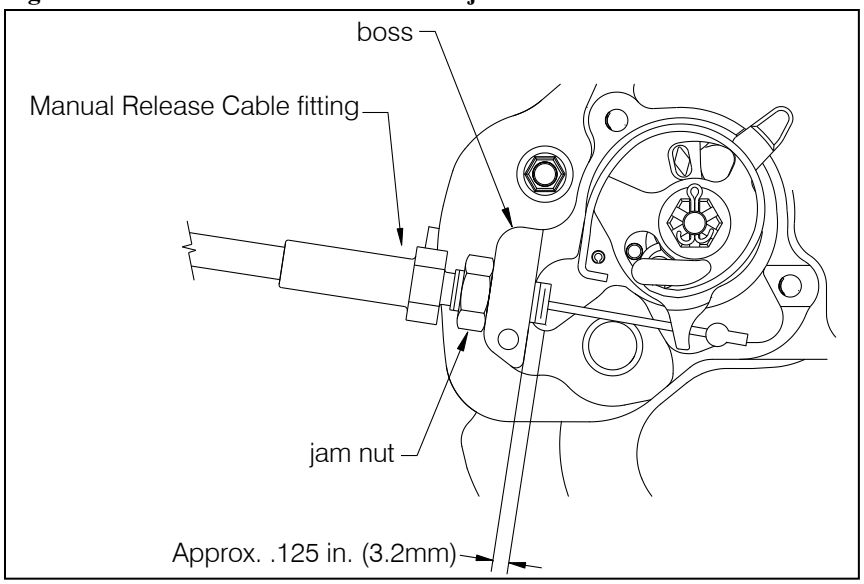
25.17 Component Re-installation continued

Manual Release Cable Installation

Connect the manual release cable (P/N 268-024-02) to the cargo hook per the following instructions:

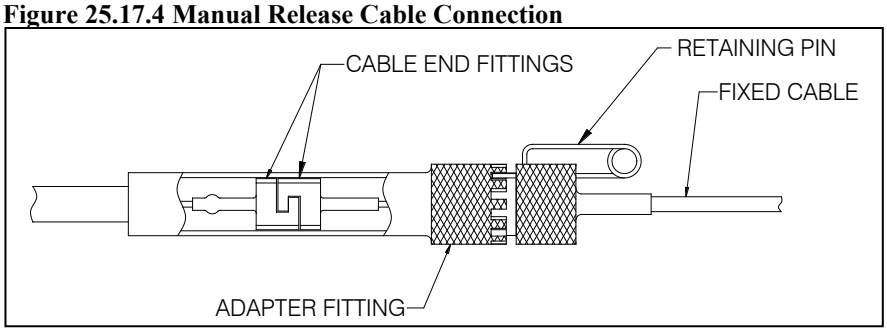
1. Remove the manual release cover from the cargo hook.
2. Thread the fitting at the end of the manual release cable into the manual release boss on the hook side plate until the threads protrude approximately .125 inches beyond the boss and secure with jam nut (as shown in Figure 25.17.3). Leave the cover off of the cargo hook until the other end of the release cable is connected, to verify proper setting.

Figure 25.17.3 Manual Release Cable Adjustment



3. Connect the other end of the release cable to the fixed section of the manual release cable by mating the cable end fittings together as shown below (slide back the Adapter Fitting to access fitting on removable cable).
4. Slide the Adapter Fitting forward and thread it onto the fixed cable fitting and engage a castellation on the Adapter Fitting with the retaining pin and lock it in place.
5. Snap the Adapter Fitting into the existing clip mounted to the belly of the helicopter.

Figure 25.17.4 Manual Release Cable Connection



25.17 Component Re-installation continued

Manual Release Cable Installation continued

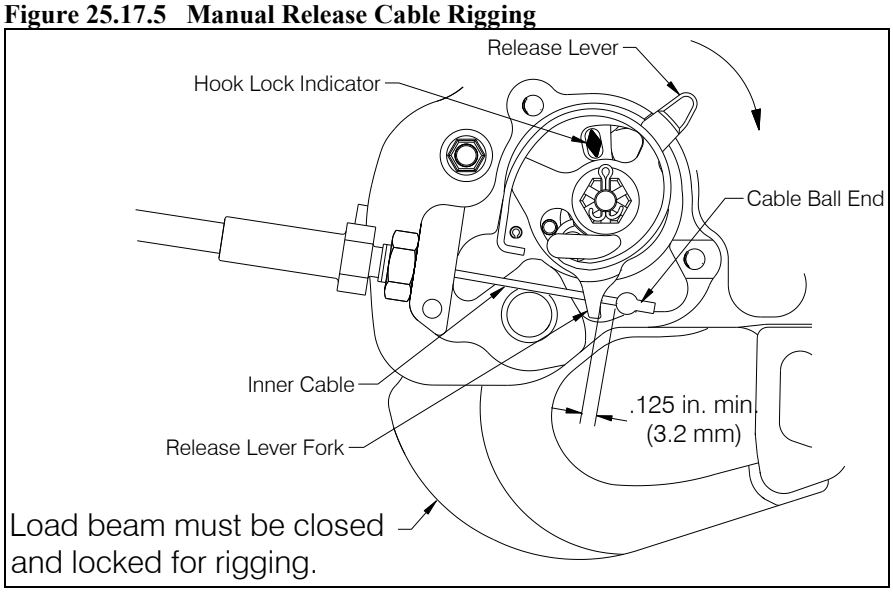
6. At the cargo hook, place the cable ball end fitting into the manual release lever fork as illustrated in Figure 25.17.5.

⚠ CAUTION

Manual release cable rigging must be done with the cargo hook in the closed and locked position.

7. With the cargo hook in the closed and locked position, rotate the release lever in the clockwise direction to remove free play (the free play is taken up when the hook lock indicator begins to move) and measure the cable ball end free play with the release lever in the cockpit in the non-release position. There must be a minimum of .125 inches (3.2 mm) between the cable ball end and fork fitting as shown in Figure 25.17.5. The maximum amount of free play is limited by the manual release cover, i.e. – the ball end must fit inside the cover when it is installed.
8. If necessary, adjust the manual release cable system to obtain a minimum of .125 inches (3.2 mm). Some adjustment can be made at the cargo hook by loosening the jam nut and turning the manual release cable or cargo hook in the required direction and re-tightening the jam nut. Ensure the manual release cable fitting threads maintain full thread engagement with the cargo hook side plate boss (i.e.- the end of the threads should not be recessed within the boss). Tighten jam nut.
9. Re-install the manual release cover with the two screws.

Figure 25.17.5 Manual Release Cable Rigging



10. Check the operation of the manual release system by pulling the release lever in the cockpit. The cargo hook must release. Reset the cargo hook by hand after release. Verify that the hook lock indicator on the side of the hook returns to the fully locked position.

25.17 Component Re-installation continued

Fixed Manual Release Cable Assembly Re-installation

1. Feed the end of the cable through the slot in the floor and re-install grommet.
2. Install the release lever assembly onto the collective stick with the two screws (P/N 510-390-00).
3. Install the cushioned loop clamp around the release cable at the bracket at frame at 1790.15 with hardware as shown in Figure 25.16.6.
4. Install the cushioned loop clamp around the release cable at the Attachment Bracket with hardware as shown in Figure 25.16.5.
5. Clip the end of the cable assembly onto the bracket on the aircraft.

Load Weigh Indicator Re-installation

1. Place the Load Weigh Indicator into the mounting bracket on the RH door pillar and secure with four screws (P/N 510-457-00).
2. Connect the electrical connector on the wiring harness to the connector on the back of the indicator.

Load Cell Re-installation

1. Attach the load cell assembly (P/N 210-249-03, 210-249-00, P/N 210-199-01, or P/N 210-199-00) to the gimbal fitting on the suspension frame with hardware as illustrated in Figure 25.16.9.
2. Tighten the nut to 60-70 in-lb and rotate to next castellation if necessary to insert cotter pin. Verify the load cell pivots independently of the bolt.

CAUTION

The Load Cell Assembly must pivot freely about its upper attach point independently of the bolt (P/N 510-443-00) and nut, if necessary, back the nut off to previous castellation to achieve this.

3. Connect the connector on the load cell harness to the connector on the belly of the helicopter.

NOTICE

A Link Assembly (P/N 232-346-00) can be installed in place of the Load Cell Assembly. It is installed using the same hardware as the Load Cell Assembly.

Self-Lubricated Bushing Re-installation

1. Install bushings with wet zinc chromate primer on the inside diameter of the mating hole.
2. Use an arbor press and an appropriately sized press tool to push the bushing into the hole until it is fully seated.

25.17 Component Re-installation continued

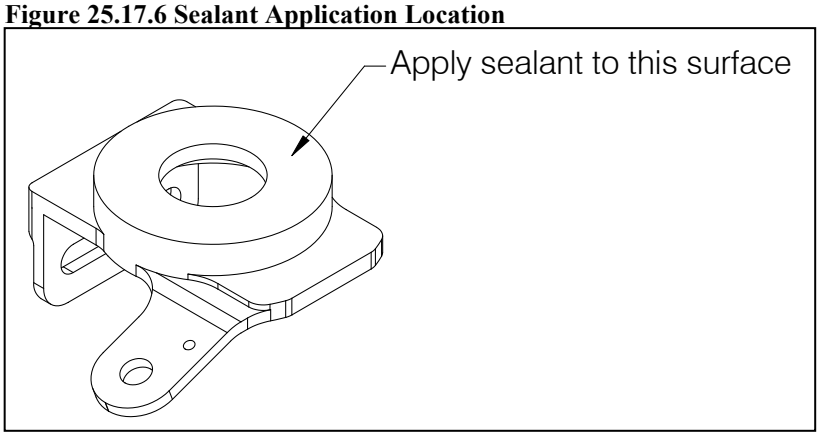
Fuel Drain Guard Re-installation

NOTICE

The fuel drain guard may not be present with all cargo swing installations (it was not included with earlier kits (prior to 2010)) and is not installed with the Airbus crash resistant fuel tank.

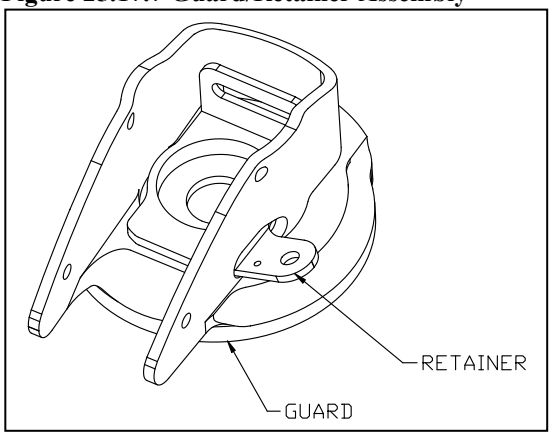
1. Remove residual sealant from the tank and Retainer, P/N 290-888-00, taking care not to mar the sealing surfaces. Prepare the areas for sealing per Airbus Helicopters Standard Practices Manual.
2. Prepare PR1422-B or equivalent sealant per Airbus Helicopters Standard Practices Manual. Apply sealant to Retainer as shown in Figure 25.17.6.

Figure 25.17.6 Sealant Application Location



3. Place the Retainer (P/N 290-888-00) inside the Guard (P/N 290-889-00), by inserting the tab through the slot in the Guard. See Figure 25.17.7.

Figure 25.17.7 Guard/Retainer Assembly



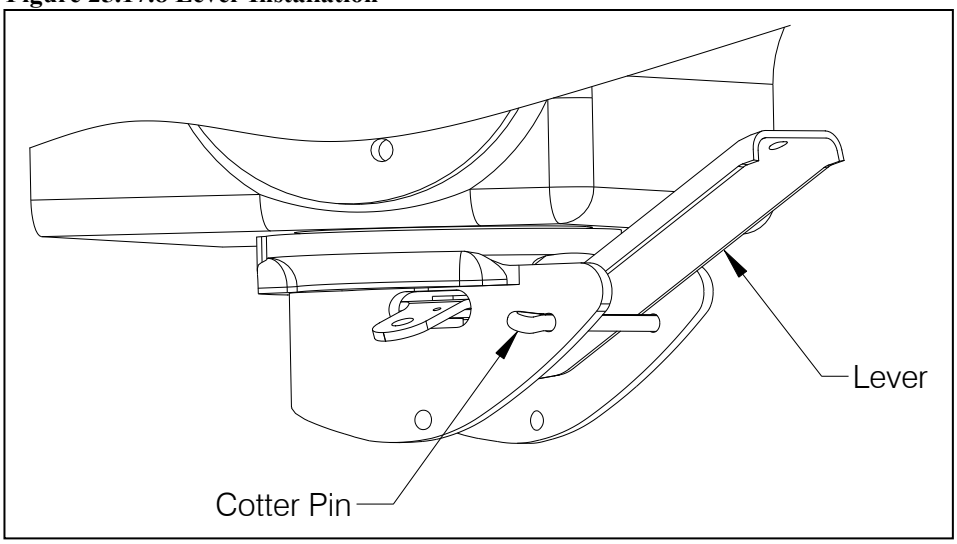
4. Place these two items onto the boss on the bottom of fuel tank, aligning the notch in the Guard with the key on the bottom of the fuel tank.

25.17 Component Re-installation continued

Fuel Drain Guard Re-installation continued

5. Secure the Guard and Retainer by re-installing the Fuel Drain Valve with new Fuel Valve Seal, 610-024-00. Use a flat-blade screwdriver to prevent the Retainer from twisting when tightening the Fuel Drain Valve. Torque per Airbus Helicopters specifications.
6. Secure the Fuel Drain Valve with safety wire using the small hole in the retainer tab.
7. Re-install the electrical connections to the Retainer tab per Airbus Helicopters Electrical Bonding Procedure. Refer to Airbus Helicopters Standard Practices Manual, 20.02.07.
8. Install the Lever by placing it in Retainer slot and rotating upwards. Secure with cotter pin (P/N 510-526-00). See Figure 25.17.8.

Figure 25.17.8 Lever Installation



9. Install a second cotter pin through the other holes in the Guard (this cotter pin is for valve protection only and is not used for rigging purposes).

25.17 Component Re-installation continued

Fuel Drain Guard Re-installation continued

10. Prepare to install Bracket (P/N 290-893-00) by threading the control cable through the Bracket hole. Install Bracket using the two screws removed previously.

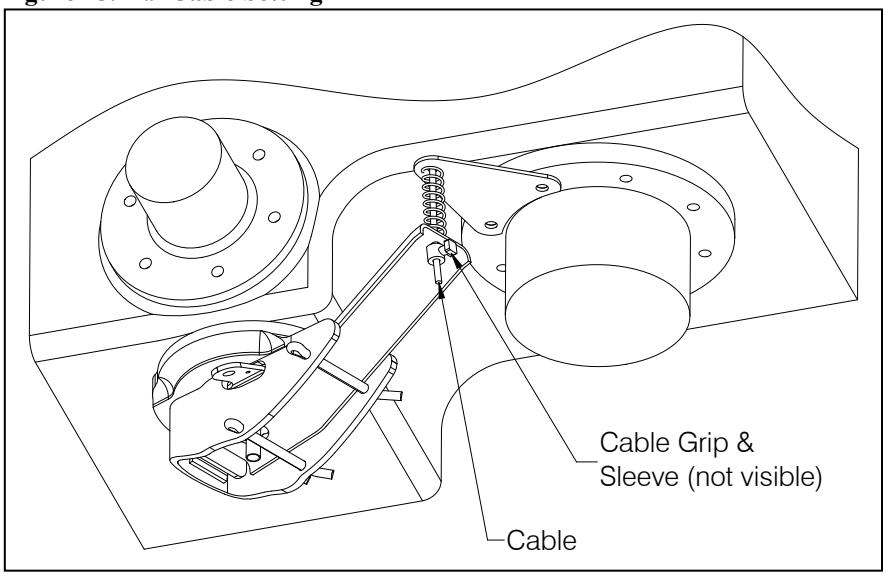
11. Pass the cable through the Spring (Airbus Helicopters P/N 350A55-1044-21) and Lever (Airbus Helicopters P/N 350A55-1043-20). Slide the Sleeve (Airbus Helicopters P/N N1-5ALU) over the cable and secure with Cable Grip (Airbus Helicopters P/N 58-2-009).

⚠ CAUTION

To avoid inadvertent fuel loss, Airbus Helicopters P/N 58-2-009 Cable Grip must be used with this installation.

12. Adjust the cable travel by doing the following: allow the lever to rest against the cotter pin stop. Slide the Cable Grip up to the bottom of the lever and secure (see Figure 25.17.9). Trim excess cable to within .25" of cable grip.

Figure 25.17.9 Cable Setting



13. Allow the sealant to cure per Airbus Helicopters Standard Practices Manual before adding fuel. Verify proper cure of unused sealant.

14. Re-fill fuel tank and check for leaks.

25.18 General Procedural Instructions-Testing

After re-installation of the cargo hook, perform the following:

1. Activate the electrical system and press the Cargo Hook release button to ensure the cargo hook electrical release is operating correctly. The mechanism should operate smoothly and the Cargo Hook must release. Reset the hook by hand after the release. If the hook does not release or re-latch, do not use the unit until the difficulty is resolved.

   CAUTION

   The release solenoid is intended to be energized only intermittently. Depressing the electrical release button continuously in excess of 20 seconds will cause the release solenoid to overheat, possibly causing permanent damage.

2. Activate the release lever located on the collective to test the cargo hook manual release mechanism. The mechanism should operate smoothly and the Cargo Hook must release. Reset the hook by hand after release. If the hook does not release or relatch do not use the unit until the difficulty is resolved.

3. Swing the installed Cargo Hook and the suspension to ensure that the manual release cable assembly and the electrical release cable have enough slack to allow full swing of each component without straining or damaging the cables. The cables must not be the stops that prevent the Cargo Hook or the suspension from swinging freely in all directions.