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Replacement Cargo Hook Kit

For The

Eurocopter EC120B

Part Number 200-351-00

Owner's Manual

Owner's Manual Number 120-139-00
Revision 0
June 16, 2009

ONBOARD™ SYSTEMS INTERNATIONAL

13915 NW 3rd Court Vancouver Washington 98685 USA
Phone: 360-546-3072 Fax: 360-546-3073 Toll Free: 800-275-0883
www.OnboardSystems.com

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RECORD OF REVISIONS

Revision	*Date*	*Page(s)*	*Reason for Revision*
0	06/16/09	All	Initial Release

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CONTENTS

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Section 1
General Information

Introduction

The 200-351-00 Cargo Hook Kit is approved as a replacement for the following Cargo Hook on the Eurocopter EC120B model helicopter.

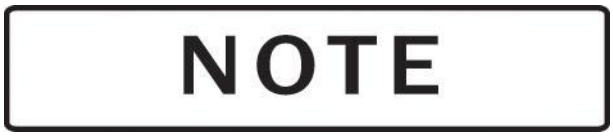
P/N	Manufacturer
AS21-8-C	Siren

Warnings, Cautions and Notes

The following definitions apply to Warnings, Cautions and Notes used in this manual.

Means that if this information is not observed, serious injury, death or immediate loss of flight safety could occur.

Means that there is a risk of injury or degradation in performance of equipment if this information is not observed.



Draws the reader's attention to information which may not be directly related to safety, but which is important or unusual.

Bill of Materials

The following items are included with the Cargo Hook Kit, if shortages are found contact the company from whom the system was purchased.

Part No.	Description	Quantity
528-029-00	3.6K Keeperless Hook	1
268-051-00	Manual Release Cable	1
290-332-00	Attach Bolt	1
410-131-00	Connector	1
510-174-00	Washer	1
510-183-00	Washer	2
510-170-00	Nut	1
510-178-00	Cotter Pin	1
120-139-00	Owner's Manual	1
121-053-00	RFMS	1
122-017-00	Cargo Hook Service Manual	1
123-034-00	ICA Manual	1

Inspection

Inspect the cargo hook for evidence of damage, corrosion and security of lock wire and fasteners. If damage is evident, do not use the unit until it has been repaired.

Specifications

Table 1-1 Specifications (528-029-00 Cargo Hook)

Design load	3,600 lb. (1,632 kg.)
Design ultimate strength	13,500 lb. (6,123 kg.)
Electrical release capacity	9,000 lb. (4,082 kg.)
Mechanical release capacity	9,000 lb. (4,082 kg.)
Force required for mechanical release at 3,600 lb.	8 lb. max. (.600” travel)
Electrical requirements	22-32 VDC 6.9 - 10 amps
Minimum release load	0 pounds
Unit weight	3.0 pounds (1.4 kg.)
Mating electrical connector	PC06P8-2S

Theory of Operation

The primary elements of the Cargo Hook are the load beam, the internal mechanism, and a DC solenoid. The load beam supports the load and is latched through the internal mechanism. The DC solenoid and an external manual release cable provide the means for unlatching the load beam.

The load is attached to the load beam by passing the cargo sling ring into the throat of the load beam and pushing the ring against the upper portion of the load beam throat, which will initiate the hook to close. In the closed position, a latch engages the load beam and latches it in this position.

To release the load, the latch is disengaged from the load beam. With the latch disengaged, the weight of the load causes the load beam to swing to its open position, and the cargo sling slides off the load beam. The load beam then remains in the open position awaiting the next load.

A load release can be initiated by three different methods. Normal release is achieved by pilot actuation of the push-button switch in the cockpit. When the push-button switch is pressed, it energizes the DC solenoid in the Cargo Hook, and the solenoid opens the latch in the internal mechanism. In an emergency, release can be achieved by operating a mechanical release cable. The release cable operates the internal mechanism of the Cargo Hook to unlatch the load beam. The load can also be released by the actuation of a lever located on the side of the Cargo Hook.

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Section 2
Installation Instructions

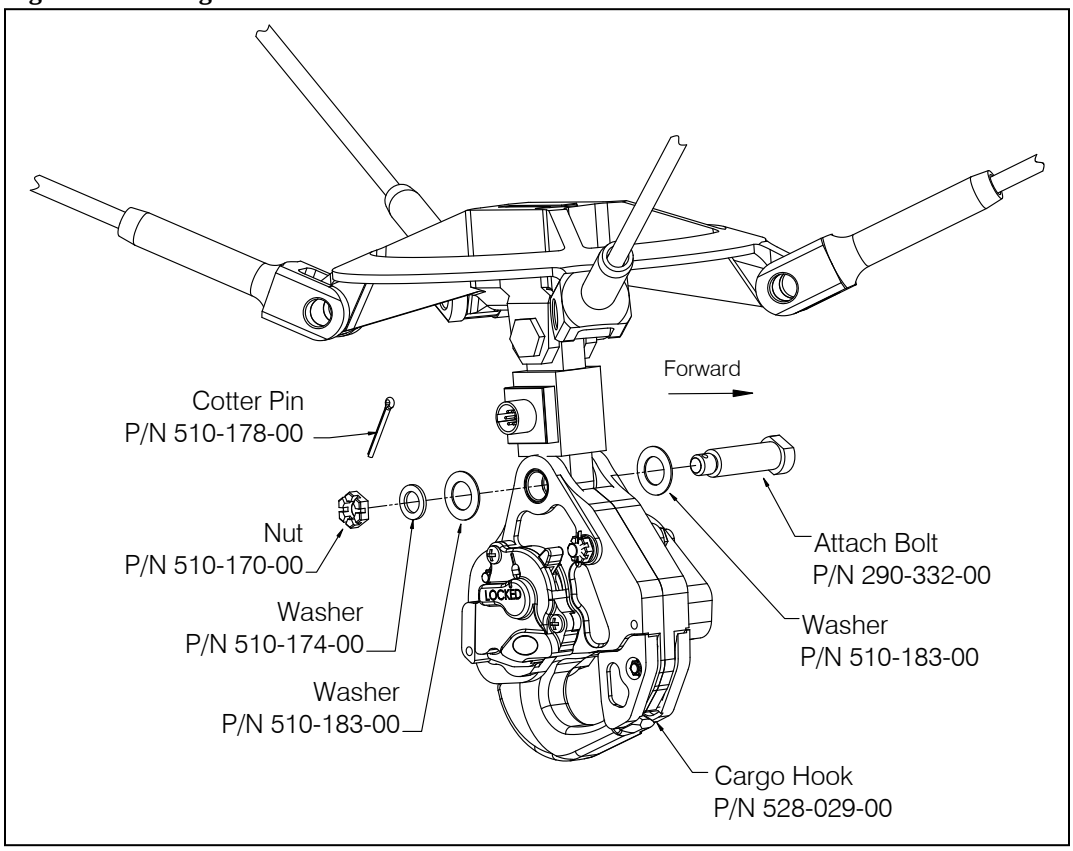
These procedures are provided for the benefit of experienced aircraft maintenance facilities capable of carrying out the procedures. They must not be attempted by those lacking the necessary expertise.

2.1 Cargo Hook Installation

Remove the existing cargo hook from the aircraft by disconnecting the cargo hook from the load cell and the manual and electrical release cables. Remove the existing external manual release cable.

Attach the new cargo hook (P/N 528-029-00) to the load cell using the hardware supplied, as illustrated below. The cargo hook load beam must point forward.

Figure 2.1.1 Cargo Hook Installation

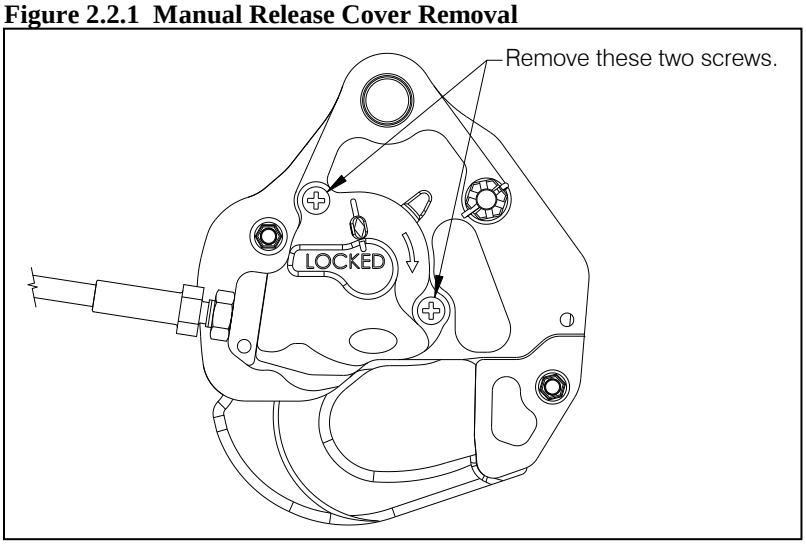


2.2 Manual Release Cable Installation

Connect the manual release cable (P/N 268-051-00) to the cargo hook per the following instructions:

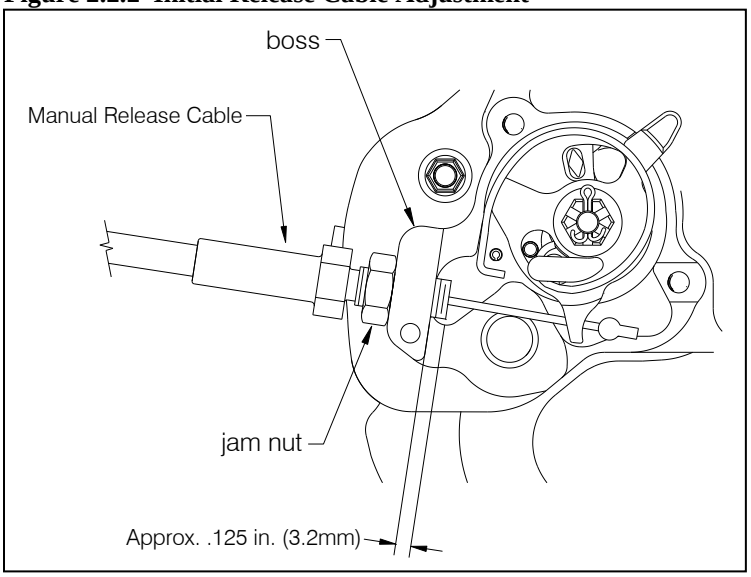
- Remove the manual release cover from the cargo hook by removing two screws (see below).

Figure 2.2.1 Manual Release Cover Removal



- Thread the fitting at the end of the manual release cable into the manual release boss on the cargo hook side plate until the threads protrude approximately .125 inches beyond the boss and secure with jam nut (as shown in Figure 2.2.2). Leave the manual release cover off of the cargo hook until the other end of the release cable is connected, in order to verify proper setting.

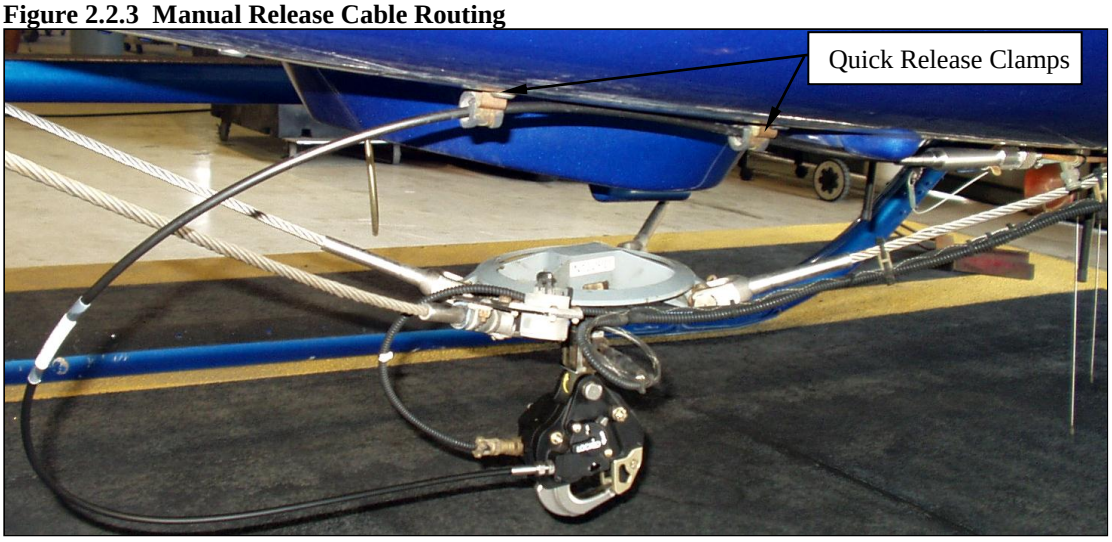
Figure 2.2.2 Initial Release Cable Adjustment



2.2 Manual Release Cable Installation continued

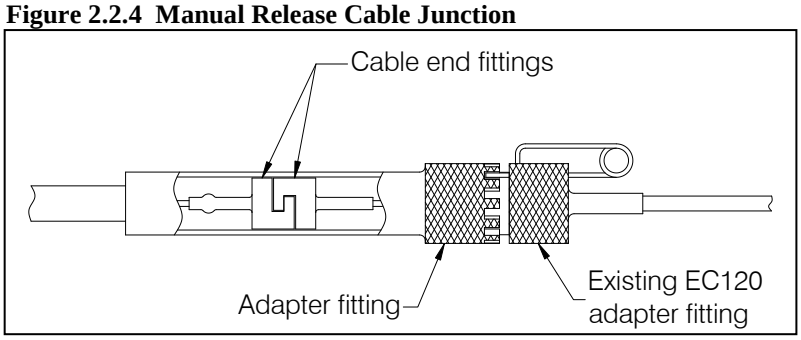
- Route the manual release cable from the cargo hook as shown below, picking up the two quick release clamps on the belly.

Figure 2.2.3 Manual Release Cable Routing



- Connect the other end of the manual release cable to the fixed section of the existing EC120 manual release cable by mating the cable end fittings together as shown below (slide back the Adapter Fitting to access the fitting on removable cable). Slide the Adapter Fitting forward and thread it onto the existing fitting, and engage a castellation on the Adapter Fitting with the retaining pin to lock it in place.

Figure 2.2.4 Manual Release Cable Junction



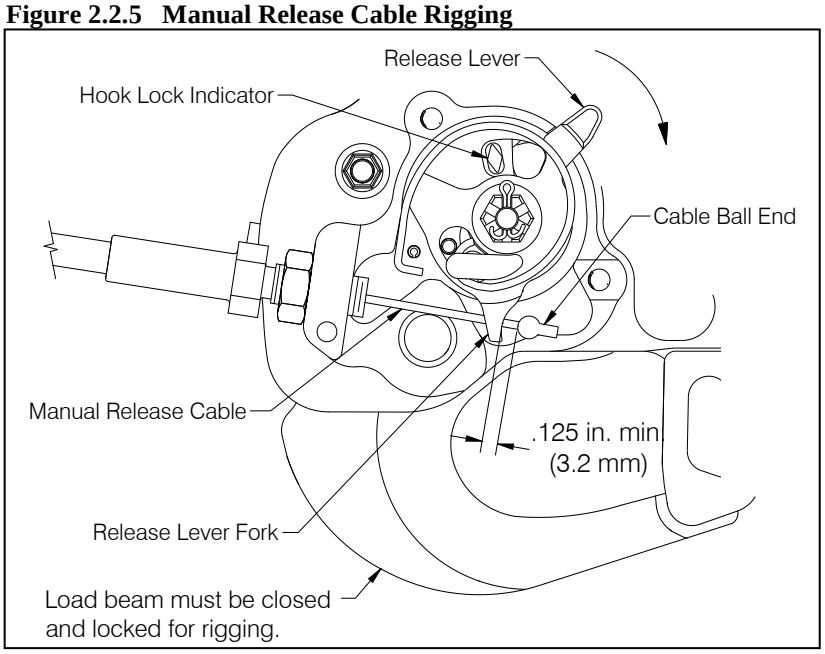
2.2 Manual Release Cable Installation, continued

- At the cargo hook, ensure the manual release cable is between the two prongs of the release lever fork as illustrated in Figure 2.2.5.

Manual release cable rigging must be done with the cargo hook in the closed and locked position.

- With the cargo hook closed and locked, rotate the release lever in the clockwise direction to remove free play (the free play is taken up when the hook lock indicator begins to move, this is also felt as the lever rotates relatively easily for several degrees as the free play is taken up) and measure the gap between the cable ball end and the release lever fork with the manual release lever in the cockpit in the non-release position. This gap should be a minimum of .125 inches (3.2 mm) as shown in Figure 2.2.5.

Figure 2.2.5 Manual Release Cable Rigging



2.2 Manual Release Cable Installation, continued

- If necessary adjust the manual release cable system to obtain the minimum gap of .125 inches at the release lever fork as shown in Figure 2.2.3 (the maximum gap is limited by the manual release cover, i.e.- the release cable must fit within the cover when it is installed). The system can be adjusted at the manual release lever on the collective or minor adjustments can be made at the cargo hook by loosening the jam nut and turning the manual release cable in the required direction (this requires that the manual release cable be disconnected from the fixed release cable and the quick release clamps on the belly). Be sure to maintain full thread engagement between the manual release cable fitting and cargo hook.
- Re-install the manual release cover with the two screws and ensure the manual release cable jam nut is tightened securely against the cargo hook.

2.3 Electrical Connector Installation

The cargo hook kit includes an electrical connector (P/N 410-131-00) to be spliced into the existing Eurocopter harness.

Cut off the existing electrical connector and splice in the supplied connector. Connect the wire from pin M at Eurocopter connector 205VC to pin A at the cargo hook and the wire from pin L at 205VC to pin B. Table 2.3.1 below shows the cargo hook connector pin out.

Table 2.3.1 Cargo Hook Connector

Pin	*Function*
A	Ground
B	Power

The cargo hook is not equipped with a hook open switch. If a third wire is present at the cargo hook, cap and stow it.

The cargo hook is equipped with a suppression diode that will be damaged if the cargo hook electrical connection is reversed. Do not attach the electrical connector until the polarity of the aircraft connector is verified to be compatible with the cargo hook connector listed in Table 2.3.1.

2.4 Installation Check-Out

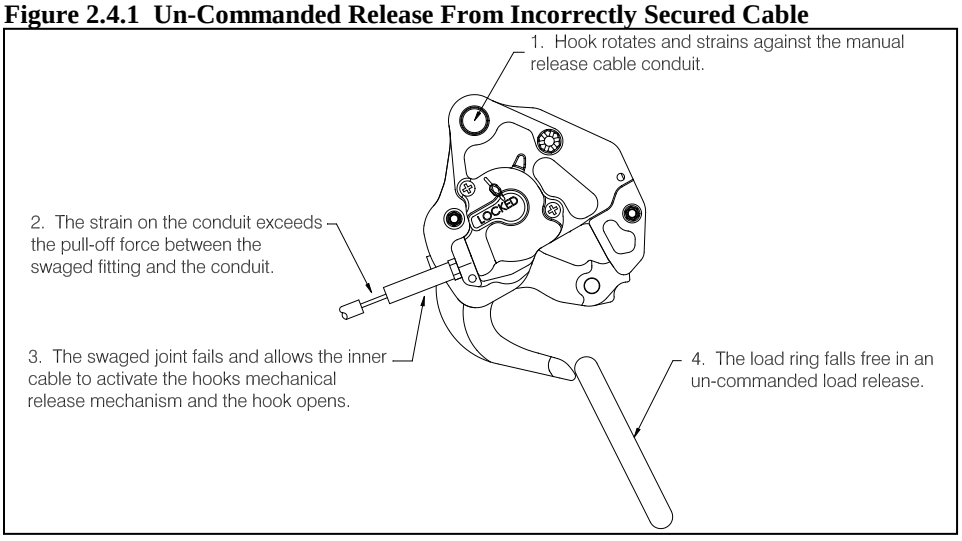
After installation of the Cargo Hook, perform the following functional checks.

1. Swing the installed Cargo Hook to ensure that the manual release cable and the electrical harness have enough slack to allow full movement of the cargo hook without straining or damaging the manual release cables or electrical harness. The harness or cable must not be the stops that prevent the Cargo Hook from swinging freely in all directions.

⚠ CAUTION

Un-commanded cargo hook release will happen if the manual release cable is improperly restrained. The manual release cable must not be the stops that prevent the Cargo Hook from swinging freely in all directions. If the Cargo Hook loads cause the hook to strain against the manual release cable the swaged end of the cable may separate allowing the inner cable to activate the cargo hook manual release mechanism. The result is an un-commanded release. Ensure that no combination of cyclic stick or cargo hook position is restrained by the manual release cable.

Figure 2.4.1 Un-Commanded Release From Incorrectly Secured Cable



2. With no load on the cargo hook load beam pull the cargo hook mechanical release lever on the collective, the Cargo Hook must release. Reset the cargo hook load beam.

3. Power up the aircraft electrical system. With no load on the cargo hook load beam, depress the cargo hook electrical release button, the Cargo Hook must release. Reset the cargo hook load beam.

2.5 Component Weights

The weights of the Cargo Hook Kit and selected components are listed below. Remember to subtract the weights of the components that this kit replaces.

Table 2.5.1 Component Weights

Item	Weight
Cargo Hook Kit	3.75 lbs (1.70 kgs)
Cargo Hook (w/o attach hardware)	3.00 lbs (1.36 kgs)

2.6 Cargo Hook Location

Refer to the Eurocopter Flight Manual Supplement for external load weight and balance data.

2.7 Paper Work

In the US, fill in FAA form 337 for the initial installation. This procedure may vary in different countries. Make the appropriate aircraft log book entry. Insert the Rotorcraft Flight Manual Supplement P/N 121-053-00 into the Rotorcraft Flight Manual.

Section 3
Operation Instructions
Operating Procedures

Prior to each job perform the following:

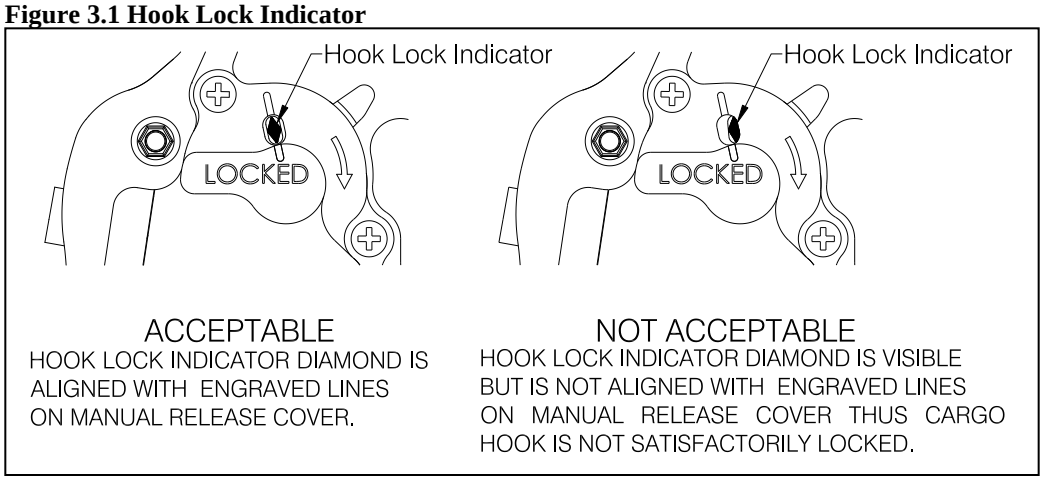
1. Ensure that the manual release cable and electrical release harness do not limit the movement of the cargo hook.
2. Be completely familiar with this manual, particularly the Cargo Hook rigging section.
3. Be completely familiar with all Eurocopter Cargo Hook operating instructions and the ICA Maintenance Manual 123-034-00.
4. Activate the electrical system and press the cargo hook release button to ensure the cargo hook electrical release is operating correctly. The Cargo Hook must release. Reset the hook by hand after release. If the hook does not re-latch do not use the unit until the difficulty is resolved.

The cargo hook release solenoid is intended to be energized only intermittently. Depressing the electrical release button continuously in excess of 20 sec. will cause the solenoid to overheat, possibly causing permanent damage.

5. Activate the manual release lever to test the cargo hook manual release mechanism. The mechanism should operate smoothly and the Cargo Hook must release. Reset the load beam by hand after release. Verify that the hook lock indicator on the side of the hook returns to the fully locked position. If the hook does not release or re-latch, do not use the unit until the problem is resolved.

In the fully locked position the hook lock indicator must align with the lines on the manual release cover (see Figure 3.1).

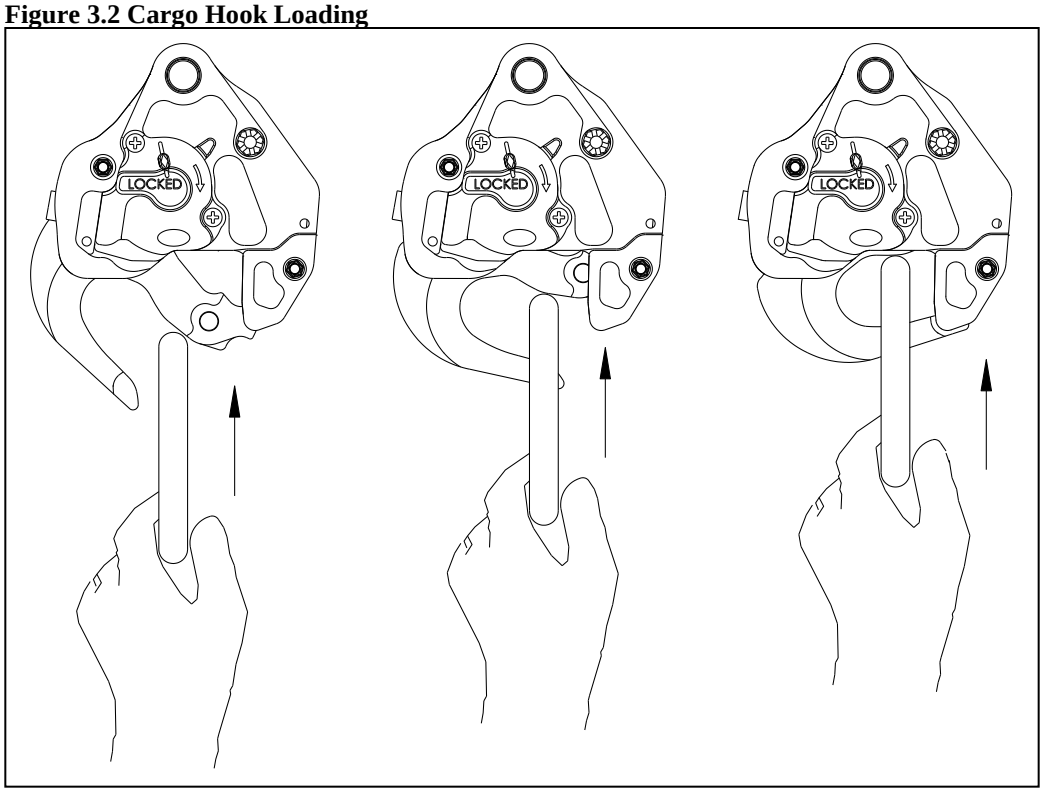
Figure 3.1 Hook Lock Indicator



Cargo Hook Loading

The cargo hook can easily be loaded with one hand. A load is attached to the hook by pushing the ring upward against the upper portion of the load beam throat, as illustrated in Figure 3.2, until an internal latch engages the load beam and latches it in the closed position.

Figure 3.2 Cargo Hook Loading



Cargo Hook Rigging

Extreme care must be exercised when rigging a load to the Cargo Hook. Steel load rings are recommended to provide consistent release performance and resistance to fouling. Figure 3.3 shows the recommended rigging, but is not intended to represent all rigging possibilities.

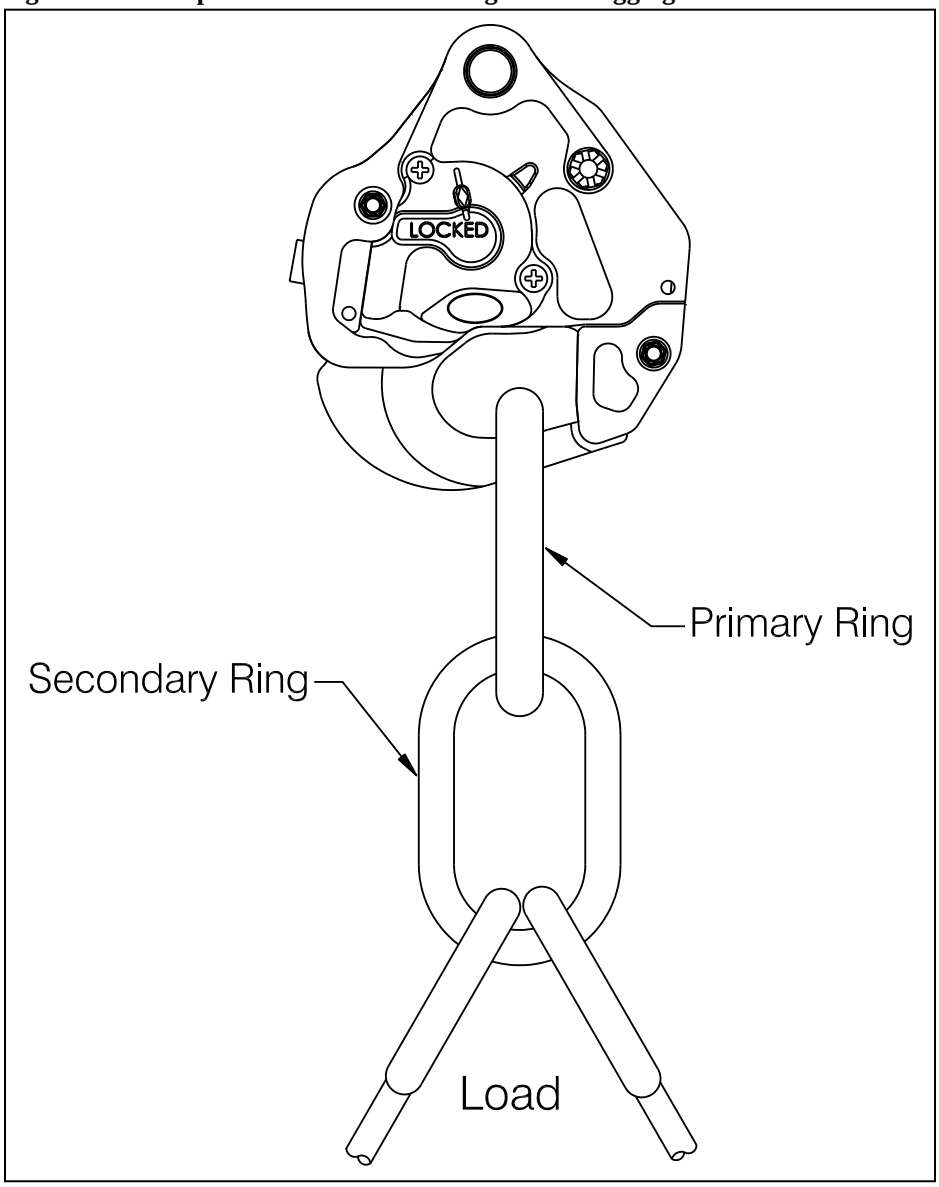
Some combinations of small primary rings and large secondary rings could cause fouling during release.

It is the responsibility of the operator to assure the cargo hook will function properly with each rigging.

Nylon type straps (or similar material) or rope must not be used directly on the cargo hook load beam. If nylon straps or rope must be used they should be first attached to a steel primary ring. Verify that the ring will freely slide off the load beam when it is opened. Only the primary ring should be in contact with the cargo hook load beam

Cargo Hook Rigging continued

Figure 3.3 Example of Recommended Cargo Hook Rigging



Section 4
Maintenance

Refer to the Instructions for Continued Airworthiness (ICA) manual 123-034-00 and Cargo Hook Service Manual 122-017-00 for maintenance.

Instructions for Returning Equipment to the Factory

If an Onboard Systems product must be returned to the factory for any reason (including returns, service, repairs, overhaul, etc) obtain an RMA number before shipping your return.

An RMA number is required for all equipment returns.

- To obtain an RMA, please use one of the listed methods.
    - Contact Technical Support by phone or e-mail (Techhelp@OnboardSystems.com).
    - Generate an RMA number at our website: http://www.onboardsystems.com/rma.php
- After you have obtained the RMA number, please be sure to:
    - Package the component carefully to ensure safe transit.
    - Write the RMA number on the outside of the box or on the mailing label.
    - Include the RMA number and reason for the return on your purchase or work order.
    - Include your name, address, phone and fax number and email (as applicable).
    - Return the components freight, cartage, insurance and customs prepaid to:

      Onboard Systems
      13915 NW 3rd Court
      Vancouver, Washington 98685
      USA
      Phone: 360-546-3072

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Section 5
Certification

STC

United States of America

Department of Transportation Federal Aviation Administration

Supplemental Type Certificate

Number **SR02028SE**

This certificate, issued to: **Onboard Systems International**
13915 NW 3rd Court
Vancouver, WA 98685

certifies that the change in the type design for the following product with the limitations and conditions therefore as specified hereon meets the airworthiness requirements of Part **27** *of the* **Federal Aviation Regulations.**

Original Product—Type Certificate Number: R00001RD
Make: Eurocopter France
Model: EC120B

Description of the Type Design Change: Fabrication of Onboard Systems International Model 200-351-00, Cargo Hook Kit, in accordance with FAA-approved Onboard Systems International Master Drawing List Number 155-144-00, Revision 0, dated July 22, 2009, or later FAA-approved revision; and installation of the 200-351-00, Cargo Hook Kit, in accordance with FAA-approved Onboard Systems International Owner's Manual Number 120-139-00, Revision 0, dated June 16, 2009, or later FAA-approved revision. This modification must be inspected and maintained in accordance with Section ATA 5 of the FAA-approved Onboard Systems International Instructions for Continued Airworthiness Document Number 123-034-00, Revision 0, dated June 2, 2009, or later FAA-approved revision and Onboard Systems International Cargo Hook Service Manual Number 122-017-00, Revision 3, dated September 17, 2008, or later FAA-approved revision.

Limitations and Conditions: Approval of this change in type design applies only to those Eurocopter model rotorcraft listed above. This approval should not be extended to other rotorcraft of these models on which other previously approved modifications are incorporated, unless it is determined by the installer that the relationship between this change and any of those other previously approved modifications, including changes in type design, will introduce no adverse effect upon the airworthiness of that rotorcraft. Rotorcraft modified in accordance with this STC must be operated in accordance with an FAA-approved copy of Onboard Systems International Rotorcraft Flight Manual Supplement (RFMS) Number 121-053-00, Revision 0, dated November 16, 2009, or later FAA-approved revision. A copy of this certificate, FAA-approved RFMS, and Maintenance Manual must be maintained as part of the permanent records of the modified rotorcraft.

If the holder agrees to permit another person to use this certificate to alter the product, the holder shall give the other person written evidence of that permission.

This certificate and the supporting data which is the basis for approval shall remain in effect until surrendered, suspended, revoked, or a termination date is otherwise established by the Administrator of the Federal Aviation Administration.

Date of application: June 12, 2009
Date of issuance: November 30, 2009
Date reissued:
Date amended:

By direction of the Administrator

(Signature)
Acting Manager, Seattle Aircraft Certification Office
(Title)

Any alteration of this certificate is punishable by a fine of not exceeding $1,000, or imprisonment not exceeding 3 years, or both.

This certificate may be transferred in accordance with FAR 21.47.

FAA Form 8110-2-1 (10-69)

EASA STC

European Aviation Safety Agency

SUPPLEMENTAL TYPE CERTIFICATE

10028873

This Supplemental Type Certificate is issued by EASA, acting in accordance with Regulation (EC) No. 216/2008 on behalf of the European Community, its Member States and of the European third countries that participate in the activities of EASA under Article 66 of that Regulation and in accordance with Commission Regulation (EC) No. 1702/2003 to

ONBOARD SYSTEMS INT.
13915 NW 3rd COURT
VANCOUVER WA 98685
USA

and certifies that the change in the type design for the product listed below with the limitations and conditions specified meets the applicable Type Certification Basis and environmental protection requirements when operated within the conditions and limitations specified below:

Original Product TC Number:	**DGAC TC NO 189**
TC Holder:	**EUROCOPTER**
Model:	**EC 120 B**
Original STC Number:	**FAA STC SR02028SE**

EASA Certification Basis:
As specified in TCDS DGAC No 189
The Certification Basis for the original product remains applicable to this certificate/ approval
The certificated noise and/ or emissions levels of the original product are unchanged and remain applicable to this certificate/ approval

Description of Design Change:
Cargo hook kit Model 200-351-00

Associated Technical Documentation:
Definition and Installation:
- Onboard Systems Master Drawing List No. 155-144-00, Rev. 0, dated July 22, 2009
- Onboard Systems Owner's Manual No. 120-139-00, Rev. 0, dated June 16, 2009

See Continuation Sheet(s)

For the European Aviation Safety Agency,

Date of issue: 16.02.2010

Massimo MAZZOLETTI
Certification Manager
Rotorcraft, Balloons, Airships

Note:
The following numbers are listed on the certificate:
EASA current Project Number: 0010003413-001

European Aviation Safety Agency

Inspection and Maintenance:
- Onboard Systems Instructions for Continued Airworthiness No. 123-034-00, Rev. 0, dated June 2, 2009
- Onboard Systems Service Manual No. 122-017-00, Rev. 3, dated September 17, 2008

Operations:
- Onboard Systems RFMS No. 121-053-00, Rev. 0, dated November 16, 2009

or later revisions of the above listed documents approved by EASA in accordance with EASA ED Decision 2004/04/CF (or subsequent revisions of this decision)

Limitations:
EASA-approved EC 120 B Flight Manual External Load Transport "Cargo Sling" FM Supplement is required.

Conditions:
Prior to installation of this modification it must be determined that the interrelationship between this modification and any other previously installed modification and/ or repair will introduce no adverse effect upon the airworthiness of the product.

This Certificate shall remain valid unless otherwise surrendered or revoked.

- end -

Note:
The following numbers are listed on the certificate:
EASA current Project Number: 0010003413-001

SUPPLEMENTAL TYPE CERTIFICATE - 10028873 - ONBOARD SYSTEMS INT.

EASA Form 91, Issue 3 - 11/11/2009

2/2

Transport Canada Approval

Transport Canada
Civil Aviation

Transports Canada
Aviation Civile

Suite 620
800 Burrard Street
Vancouver, B.C.
V6Z 2J8

Your file Votre référence

Our file Notre référence
NAPA# P-10-0034
RDIMS 5793109

April 15, 2010

Mr. Mark Hanson
Onboard Systems International
13915 NW 3rd Court
Vancouver, WA 98685
USA

Subject: Acceptance of Foreign STC SR02028SE

Dear Mr. Hanson,

This is in response to FAA letter dated December 24, 2009, Reference 130S-GA-09-096, requesting Transport Canada approval of the subject STC.

In accordance with our current policy associated with the review of foreign STCs, some STCs applicable to certain categories of aircraft may be accepted solely on the basis of their foreign certification, and do not require the issue of a corresponding certificate by Transport Canada. The subject STC falls within these criteria.

This STC will be entered in the national index of STCs that have been reviewed and accepted by Transport Canada for installation on Canadian-registered aeronautical products.

This letter confirms formal acceptance of the referenced STC by Transport Canada.

Yours truly,

Henry Wong
for Regional Manager
Aircraft Certification

c.c.: Manager, Seattle Aircraft Certification Office

Canada

1/1